

Model 100 Series Piston Engine Continued Airworthiness Program

TEMPORARY REVISION NUMBER 2

DATED 7 January 2000

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This Temporary Revision consists of the following pages, which affect and replace existing pages in the paper copy manual and supersede aerofiche information.

SECTION PAGE		AEROFICHE FICHE/FRAME	SECTION	PAGE	AEROFICHE FICHE/FRAME
					,
Intro 57-50-00	i 3-91	1-A03 1-F09	57-60-00 57-60-00	3-97 3-98	1-F15 1-F16
57-50-00	3-92	1-F10			

REASON FOR TEMPORARY REVISION

To update inspection procedures, time of compliance, affected airplane models and serial numbers.

FILING INSTRUCTIONS FOR THIS TEMPORARY REVISION

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TEMPORARY REVISION NUMBER 1

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х	1-A12	55-10-02	3-57	1-D21
xi	1-A13	57-10-03	3-83	1-F01
xii	Added	57-10-05	3-85	1-F03
	PAGE i x xi xi	i 1-A03 x 1-A12 xi 1-A13 xii Added	AEROFICHE FICHE/FRAME SECTION i 1-A03 53-40-01 x 1-A12 55-10-02 xi 1-A13 57-10-03 xii Added 57-10-05	AEROFICHE FICHE/FRAME SECTION i 1-A03 53-40-01 3-51 x 1-A12 55-10-02 3-57 xi 1-A13 57-10-03 3-83 xii Added 57-10-05 3-85

REASON FOR TEMPORARY REVISION

To update inspection procdures and equipment source.

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CONTINUED AIRWORTHINESS PROGRAM

MODEL 100 SERIES PISTON SINGLE ENGINE AIRCRAFT



THIS REISSUE SUPERSEDES AND REPLACES CONTINUED AIRWORTHINESS PROGRAM MANUAL D5130-13 DATED 3 OCTOBER 1994.

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3 APRIL 1995

D5133-13

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APPLICABILITY

MODEL

SERIAL

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150K Thru 150M 152 A152	I hru 15079405 A1500001 Thru A1500734 15279406 Thru 15286033 681, A1500433, A1520735 Thru A1521049
170 Thru 170B 172 Thru 172Q	609, 1800 Thru 27169 610, 612, 615, 622, 625, 630, 638, 639, 2800 Thru 17276211
P172D R172E Thru 172K	P17257120 Thru P17257188 680, R172-0001 Thru R1723454
172RG	691, 172RG0001 Thru 172RG1191
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182 Thru 182R	613, 631, 634, 675, 33000 Thru 18268586
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188 Thru 188C	653, 678T, 188-0001 Thru T18803974T
190/195/195A/195B	7004 Thru 16183

OBJECTIVE

1.0. DISCUSSION.

The Continued Airworthiness Program (CAP) consists of inspection recommendations that, when combined with existing maintenance requirements, will help maintain the continued airworthiness of the aircraft.

1-1. INTRODUCTION.

The aircraft contained in this CAP were delivered with recommended inspection programs. Service Manuals continually kept up to date supplemented by service bulletin and owner information were provided and intended to keep the aircraft airworthy. Due to the varied missions the aircraft were assigned, type of care given, as well as the general age and utilization rate, Cessna feels it necessary to bring additional requirements to inspection programs as a means of further assuring the aircraft can continue to be airworthy. From experience we have gained from continued support of the aircraft and inputs from the field, Cessna continues to maintain service information for the aircraft through the Continued Airworthiness Program.

This CAP has been developed in accordance with guidelines given in FAA Advisory Circular 91-56 and 91-60. The CAP is not intended to change or replace, but could provide additional information to the Service Manual or Service Letters which are applicable.

1-2. OBJECTIVE.

This CAP has been developed to expand present inspection requirements. Some are more stringent to further assure the ability of the airplane to perform within the limits of the original certification.

The CAP will address primary and secondary airframe components, powerplant, electrical items and primary and secondary systems to accomplish this stated objective of continued airworthiness.

To establish the basis for these items to be included, the following assumptions have been made:

- A. The aircraft has been maintained in accordance with Cessna recommendations or equivalent.
- B. Where the CAP is directed to a specific part or component, it is implied that the inspection will include observation and evaluation of the surrounding area of parts and equipment. Any discrepancies found during this inspection outside of the scope of the CAP should be reported to Cessna through the existing condition reporting system so that changes can be made to the CAP where necessary.
- C. Aircraft modified by STC are not the responsibility of Cessna. Any inspections called for in Cessna manuals or Cessna's CAP's that have areas that have been modified by STC's shall automatically be referred to the STC holder by the owner and/or maintenance organization for obtaining FAA approval guidelines.

2-0. STRUCTURAL/SYSTEM SIGNIFICANT DETAILS (SSD'S).

The items designated for inspection in this document are those structures or systems which include powerplant and electrical whose failure could cause loss or substantial damage to the aircraft. The terms <u>Structurally Significant Details or System Significant Details (SSD's)</u> will be applied to these critical items. The selection and monitoring of these SSD's is the nucleus of the Continued Airworthiness Program.

2-1. RATIONALE USED TO SELECT SSD'S.

The Cessna 100 Series fleet was designed for business and pleasure flights. However, since some aircraft are at 10,000 flight hours and others are aged (20 + years) with low flight hours common, the methods and rationale for selecting the critical SSD's are provided in paragraph 2-2, A. thru E.

2-2. THE SSD'S WERE SELECTED BASED ON THE FOLLOWING:

A. SERVICE EXPERIENCE.

Customer correspondence and Condition Reports were also reviewed during the selection process for the critical SSD's. Reports were also used to select items that were similar in application to other aircraft based on their design and loading.

B. SERVICE BULLETINS, NEWSLETTERS, AND SERVICE KITS.

The Service Bulletins were reviewed and readdressed for safety-of-flight criteria to help select critical SSD's. Some of these documents are considered vital to the safe operation of the aircraft and, as a result, the affected structure or system component will be listed as an SSD.

NOTE

Some of these bulletins may already be incorporated into an operator's maintenance program.

C. ANALYSIS.

Existing analyses were reviewed to identify components in areas that may have exhibited the potential for additional inspection requirements.

D. TESTS.

A review of test results applicable to the design was made. The loading conditions together with the static and fatigue test results were evaluated. The resulting data were used to determine if the component should be considered as an SSD for incorporation into the CAP.

E. INSPECTION OF AIRCRAFT.

High-time aircraft were disassembled for inspection to assist in selecting SSD's. Other high-time aircraft special inspection results supplement these inspections. The effects of corrosion have not been considered in the rationale in selecting SSD's or in the effects on the fatigue stresses. It is difficult to determine the effects of corrosion, but as the reports from the CAP inspections are received, the inspection times will be adjusted as necessary dependent upon the severity of the discrepancy.

NOTE

If aircraft has any history of low level overland survey the initial and repeat inspection interval is halved.

3-0. REPORTING-COMMUNICATIONS.

For the CAP to be successful on a continuing basis, it is essential that a free flow of information exist between the operator, FAA and Cessna. The significant details of inspection results, repairs and modifications accomplished must be communicated to Cessna in order to assess the effectiveness of their recommended inspection procedures and time intervals. In some cases, extensions of inspection frequencies may be possible if the data suggests that the onset of fatigue problems occurs at a greater number of flight hours than initially predicted.

Additionally, items not previously considered for inclusion in the CAP may be uncovered through operator inspections and reporting. These items will be evaluated by Cessna and, if applicable generally to the aircraft configurations concerned, will be added to the document for the benefit of all.

A reporting system, consistent with the systems employed by CESSNA PROPELLER AIRCRAFT PRODUCT SUPPORT ORGANIZATION has been established and incorporated into this document. Copies of the appropriate forms are available to you from a Cessna Service Station or Factory Field Service Engineer.

3-1. DISCREPANCY REPORTING.

Discrepancy reporting is essential to provide for adjusting the inspection thresholds and the repeat times up or down, as well as adding or deleting SSD's. Based on the data reported, it may be possible to improve the inspection methods, repairs and modifications involving the SSD's.

All cracks and significant corrosion found involving the inspection of an SSD shall be reported to Cessna within 10 days. The SSD inspection results are to be reported on a form as shown on the following pages.

3-2. DISCREPANCY FORM DISPOSITION.

Send all available data including repairs, photos, etc., to:

Cessna Aircraft Company Attn: CAP Program Technical Support Services Dept. 751 P.O. Box 7706 Wichita, Kansas 67277

NOTE

This system does not supersede the condition reporting system of communication for items not covered by the CAP.

3-3. CESSNA FOLLOW-UP ACTION.

All SSD reports will be reviewed to see if any of the following action should be taken:

- 1. Check effect on structural or operational integrity.
- 2. Check other high-time aircraft to see if Service Bulletin should be considered.
- 3. See if reinforcement is required.
- 4. Revise CAP inspection document if required.

4-0. BASIC INFORMATION.

This CAP applies to the 100 Series Aircraft. Refer to individual 100 Series Service Manuals for specific inspection programs on your aircraft.

4-1. INSPECTION.

1. GENERAL

- A. A recommended method of inspection is given; however, if the operator can show that an alternate method is equal to or better than the suggested method for detecting damage, then it will be considered as a possible alternate inspection method.
- B. The following general eddy current inspection shall be used where called out as "INSPECTION PROCEDURES" within Section III.
- C. Eddy current inspect using an eddy current test instrument capable of detecting the simulated cracks in a bolt hole and surface reference standard equivalent to the HRS-10A and SRS-123A standard available from NDT Engineering, 7056 S. 220th, Kent, WA 98032
- D. The following general liquid penetrant inspection shall be used where called out as "INSPECTION PROCEDURES" within Section III.

E. Penetrant inspection is used to detect small cracks or discontinuities open to the surface which may not be evident by normal visual inspection. Penetrant inspection can be used on most airframe parts and assemblies accessible for its application. The inspection is performed by applying a liquid which penetrates into surface defects so that visual indications are obtained by color contrast or fluorescence of the penetrant under the display of black light. The penetrant method of inspection requires that the surface in the inspection area be thoroughly clean and free of paint.

4-2. MANUALS/PARTS.

Cessna has a number of documents that are useful to maintaining continued airworthiness of aircraft:

Cessna 100 Series Service Manuals. Cessna 100 Series Parts Catalogs Cessna Single Engine Service Bulletin Summaries Cessna Newsletter Summaries

For information regarding these documents, contact:

Cessna Parts Distribution Attn: SPA Department P.O. Box 949 Wichita, Kansas 67218

4-3. SERVICE LETTERS/BULLETINS.

As an aid to the operator, a listing of all the Service Letters/Bulletins pertaining to these 100 Series Aircraft CAPS, are listed in Section 1, titled: TECHNICAL DOCUMENT REFERENCE. For information concerning the technical data included in these Service Letters/Bulletins that apply to your aircraft, you may contact Cessna's Propeller Technical Information Services, Department 753. The telephone number for customer assistance is (316) 941-7550. A Service Bulletin Listing Program, that provides a list of all Cessna Service Bulletins and Service Newsletters, applicable to a particular airplane model and serial number is also available from Cessna. This service is obtained by calling (316) 941-6118.

5-0. APPLICABILITY/LIMITATIONS.

The Cessna 100 series aircraft have had many modifications that were accomplished under STC's by other organizations without Cessna Engineering involvement. Because Cessna does not have knowledge of the airworthiness of STC's, this report cannot apply to those areas affected by such STC's and the STC holder should prepare a CAP for each particular modification.

6-0. SIGNIFICANT DETAILS.

This section contains the significant details selected by the rationale process described in paragraph 2-2. These items are considered significant to maintain continued airworthiness of all Cessna 100 series models. Service Bulletins considered mandatory for airworthiness will be listed in Section 1, TECHNICAL DOCUMENT REFERENCE.

A summary of the CAPs is shown in the section titled: LISTING OF CONTINUED AIRWORTHINESS INSPECTIONS. This can be used as a checklist by the operators.

6-1. CAP FORMS.

Each CAP will be listed on a form and will contain the following:

- 1. Continued Airworthiness Inspection Number.
- 2. Title.
- 3. Effectivity.
- 4. Inspection Compliance.
- 5. Repeat.
- 6. Purpose.
- 7. Inspection Procedure.
- 8. Access/Location.
- 9. Detectable Crack Size.
- 10. Procedure.
- 11. Repair/Modification.
- 12. Comments.

6-2. REPAIR INFORMATION/MODIFICATION

Repairs may be made in accordance with the applicable Cessna Service/Maintenance Manual or FAA AC43-13 acceptable methods of alteration and repair. Any repair not covered by the recommendations in this CAP Program may be coordinated with Cessna Technical Information Services at Telephone (316) 941-7550.

Temporary Revision Number 1 July 15, 1998

Discrepancy Report

CAP'S NO: AIRPLANE LOCATION:			S/N OF AIRPLANE:	
INSPECTION CONDUC	TED:	Date	Airplane Total Hours	Cycles
SERVICE HISTORY:			Component Total Hours	sCycles
		·····		
		S:		
ACCESS REQUIRED:				
REPAIR DESCRIPTION:				
COMMENTS:				
Enclose all available data	a inclu C A T C	ding photos essna Airc ttn: CAP P echnical S ept. 751C	s, sketches, etc to : raft Company rogram upport Services	

P.O. Box 7706

Wichita, Kansas 67277 FAX 1-316-942-9006

SECTION I - TECHNICAL DOCUMENT REFERENCE

MANUALS

Title	Number	Model
Service Information Summary	D195-13	120 & 140 (1946 - 1949) 140A (1949 - 1951) 170 - 170B (1948 - 1956)
100 Series Sevice Manual	D138-13	150 - 150B (1959 - 1962) 172 - 172C (1956 - 1962) 175 - 175C (1958 - 1962) 182 - 182E (1956 - 1962) 185 - 185A (1961 - 1962)
100 Series Sevice Manual	D637-13	150C - 150H (1963 - 1968) 172D - 172! (1963 - 1968) 182F 182L (1963 - 1968) 185B - A185E (1963 - 1968)
150 Series Sevice Manual	D971-3-13	150J - 150M (1969 - 1976)
150 Series Sevice Manual	D2011-13	150M (1977)
152 Series Sevice Manual	D2064-13	152 (1978 - 1985)
172/Skyhawk Series Sevice Manual	D972-3-13	172K - 172M (1969 - 1976)
172/172Q Series Sevice Manual	D2065-2-13	172N and 172Q (1977 - 1986)
Hawk XP Sevice Manual	D2027-13	172P (1977 - 1981)
172RG Sevice Manual	D2066-13	172RG (1980 - 1985)
177/Cardinal Sevice Manual	D841-7-13	177 - 177B (1968 - 1978)
Cardinal RG Sevice Manual	D991-3-13	177RG (1971 - 1975)
Cardinal RG Sevice Manual	D2009-3-13	177RG (1976 - 1978)
180/185 Skywagon & AGcarryall Sevice Manual	D2008-8-13	180 - 180K (1969 - 1980)
180 - 185 Series Sevice Manual	D2067-13	182M - 182P (1969 - 1976) A185F (1981 - 1985)
182/Skylane Series Sevice Manual	D2006-3-13	180K (1981)
182/T182 Skylane Series Sevice Manual	D2068-2-13	182Q - 182R (1977 - 1986)
R182/TR182 Skylane RG Sevice Manual	D2069-2-13	R182 - TR182 (1978 - 1986)
180/185 Skywagon & AGcarryall Sevice Manual	D2000-8-13	A185E - A185F (1969 - 1980)
188 AG Aircraft Series Sevice Manual	D2054-1-13	188 - T188C
To obtain Manuals, write to:		

Cessna Parts Distribution Att: SPA Dept P.O. Box 949 Wichita, Kansas 67218

SERVICE BULLETINS

Number	Title	Date	Cap Insp
SL64-8	Control Wheel Inspection	2/14/64	27-10-00
SL65-1	Baggage Compartment Corrosion	1/5/65	53-20-00
SL67-60	Aileron End Bearing Inspection	11/21/67	27-10-01
SE70-32	Horizontal Stabilizer Attachment Brackets Inspection	12/18/70	55-10-02
SE71-23	Horizontal Stabilizer Attachment Inspection	8/6/71	71-60-00
SE71-27	Cowl Flap Hinge Pin Inspection	11/12/71	57-10-00
SE72-03	Vertical Stabilizer Attachment Inspection	2-11-72	55-30-03
SE72-29	Vertical Fin Attachments	9/29/72	53-40-00
SE73-12	Aileron Cable Inspection	5/18/73	27-10-02
SE74-14	Fuel Vent Line Inspection	6/21/74	28-20-03
SE78-62	Vertical Fin Attach Bracket Inspection	10/13/78	55-30-00
SE79-49	Vertical Fin Attach Bracket Nutplate Inspection	4/28/80	55-30-02
SE80-57	Elevator Trim Tab Inspection	5/12/80	27-30-00
SE80-65	Rudder Stop Clip Inspection	6/16/80	27-20-00
SE80-85	Vertical Fin Rear Spar Inspection	9/29/80	55-30-01
SE80-96	Oil Cooler Inspection	1/12/81	79-10-00
SE84-8	Quick Drain Valves - Fuel Reservoir and Fuel Tanks Inspection	3/16/84	28-30-01
SNL85-54	Hydraulic Hose Replacement and Hydraulic Component Replacement	11/27/85	29-10-01
SNL86-49	Main Landing Gear Actuator Mounting Bolts	12/12/86	32-30-01
SEB89-1	Stabilizer Balance Weight Bracket	12/21/90	55-20-03
SEB90-1	Main Landing Gear Pivot Inspection	9/21/90	32-10-00

Service Bulletins listed above, write to:

Cessna Aircraft Company Dept 753C P.O. Box 7706 Wichita, Kansas 67277

Continued Airworthin Program Inspectior Number	d Ness N	Inspection Compliance Title	Model	Effoctivity	Initial	Popost
			Nodel	Enectivity	Innuar	переа
27-10-00	Contr Inspe	ol Wheel ction	150 Thru 150M	617, 628, 644, 649, 17001 Thru 15079405	1000 Hrs	1000 Hrs
			A150K Thru	A1500001		
			A150M	Thru A1500734		
			A-150L	A-1501001		
				Thru A-1501039		
			A-A150L	A-A1500001		
				Inru A-A1500009		
			F150F Thru F150M	F150-0001 Inru F15001429		
			FA150K Thru	FA15000428		
			FA150M	FA1500336		
			172 Thru 172Q	610, 612, 615, 622,		
				625, 630, 638, 639,		
				2800 Thru 17276211		
			P172D	P17257120		
				Thru P17257188		
			FP172	FP172-0001 Thru		
				FP172-0003		
			F172D Inru F172D	F1/2-0001 Inru		
			FR172F Thru	FR17202204		
			FR172K	FB17200675		
			175 Thru 175C	619, 28700A.		
				55001 Thru 17557119		
			180 Thru 180K	604, 614, 624, 645,		
				30000 Thru 18053203		
			182 Thru 182R	613, 631, 634, 675,		
			D.400	33000 Thru 18268586		
			R182	R1820001		
			A192 Thru	10ru 18202039		
			Δ182N	A182-0001 Thru A182-0148		
			F182P Thru	F18200001 Thru		
			F182Q	F18200169		
			F182RG	FR18200001 Thru		
				FR18200070		
			188 Thru T188C	653, 678T,		
				188-0001		
				Thru T18803974T		
			A-A188B	A-A1880001 Thru		
				A-A188034		

Continued Airworthiness Program Inspection Inspection Compliance					
Number	Title	Model	Effectivity	Initial	Repeat
27-10-01	Aileron End Bearing Inspection	177 Thru 177B 177RG F177RG	661, 17700001 Thru 17702752 177RG0001 F177RG0001 Thru F177RG0177 Thru 177RG1366	1000 Hrs or 3 Yrs	1000 Hrs or 3 Yrs
		188 Thru T188C	653, 678T, 188-0001 Thru T18803974T		
		A-A188B	A-A1880001 Thru A-A188034		
27-10-02	Aileron Cable Inspection	188 Thru ⊺188A	653, 678T, 188-0001 Thru 1880707	500 Hrs	500 Hrs
		A-A188B	A-A1880001 Thru A-A188034		
27-20-00	Rudder Cable Inspection	152	15279406 Thru 15286033	1000 Hrs	1000 Hrs
		A152	681, A1500433, A1520735 Thru A1521049		
		F152	F15201429 Thru F15201980		
		FA152	FA1520337 Thru FA1520425		
28-20-01	Hose - Fuel Line Inspection	150 Thru 150M	617, 628, 644, 649, 17001 Thru 15079405	5 Yrs	2 Yrs
		A150K Thru	A1500001		
		A 150M A-150I	A-1501001		2 Yrs
		N 100E	Thru A-1501039		
		A-A150L	A-A1500001 Thru A-A1500009		
		F150F Thru	F150-0001 Thru		
			F15001428		
		FAISUN INTU	FA1500001 INFU		
		152	15279406 Thru 15286033		
		A152	681, A1500433, A1520735 Thru		
		F152	A1521049 F15201429 Thru		
		1102	F15201980		

SECTION II - LISTING OF CONTINUED AIRWORTHINESS PROGRAM INSPECTIONS

Continued Airworthine Program Inspection	ess	Inspection Compliance				
Number		Title	Model	Effectivity	Initial	Repeat
28-20-01	Hose Insp	e - Fuel Line ection	FA152	FA1520337 Thru FA1520425	5 Yrs	2 Yrs
			170 Thru 170B	609, 1800 Thru 27169		
			172 Thru 172Q	610, 612, 615, 622, 625, 630, 638, 639, 2800 Thru 17276211		
			P172D	P17257120		
				Thru P17257188		
			B172E Thru	680, B172-0001		
			172K	Thru B1723454		
			172RG	691, 172RG0001		
				Thru 172RG1191		
			FP172	FP172-0001 Thru		
				FP172-0003		
			F172D Thru	F172-0001 Thru		
			F172P	F17202254		
			FR172E Thru	FR17200001 Thru		
			FR172K	FR17200675		
			175 Thru 175C	619, 28700A,		
				55001 Thru 17557119		
			177 Thru 177B	661, 17700001 Thru 17702752		
			177RG	177RG0001 Thru 177RG1366		
			F177 RG	F177RG0001 Thru F177RG0177		
			180 Thru 180K	604, 614, 624, 645, 30000 Thru 18053203		
			182 Thru 182R	613, 631, 634, 675,		
				33000 Thru 18268586		
			R182	R1820001		
				Thru 18202039		
			A182J Thru	A182-0001 Thru		
			A182N	A182-0148		
			F182P Thru	F18200001 Thru		
			F182Q	F18200169		
			F 182HG	FR1820001 Ihru		
			188 Thru T1000	FR102000/0		
				198-0001		
				Thru T1880207/T		
			A-A188R	Δ-Δ1880001 Τμμ		
				A-A188034		
			190/195/	7004 Thru 16183		
			195A/195B			
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Continued Airworthine Program Inspection Number	ess Inspection Compliance Title	Model	Effectivity	Initial	Repeat
28-20-02	Hose Inspection	180 Thru 180K	604, 614, 624, 645,	5 Yrs	2 Yrs
	- Fuel Vent	182 Thru 182D	30000 Thru 18053203 613, 631, 634, 675, 33000 Thru B18202041		
		A182J Thru A182N E182D Thru	A182-0001 Thru A182-0148		
		F182Q 185 Thru A185F	F18200001 1110 F18200169 632, 652, 185-0001 Thru 18504448		
28-20-03	Fuel Vent Line	177 Thru 177B	661, 17701974	1000 Hrs	1000 Hrs
	Inspection	177RG	177RG0433 Thru 177RG0592		
		F177RG	F177RG0001 Thru F177RG0177		Hrs 1000 Hrs ⁄rs 2Yrs
28-30-01	Quick Drain Valves - Fuel Reservoir and	150 Thru 150M	617, 628, 644, 649, 17001 Thru 15079405	1000 Hrs or 2 Yrs	1000 Hrs 2Yrs
	Fuel Tank Inspection	A150K Thru A150M A-150L	A1500001 Thru A1500734 A-1501001		
		A-A150L	Thru A-1501039 A-A1500001 Thru A-A1500009		
		F150F Thru F150M	F150-0001 Thru F15001428		i 1000 Hrs 2Yrs
		FA150K Thru	FA1500001 Thru		
		152	15279406 Thru 15286033		
		A152	681, A1500433, A1520735		
		170 Thru 170B 172 Thru 172Q	1 hru A1521049 609, 1800 Thru 27169 610, 612, 615, 622, 625, 630, 638, 639, 2800 Thru 17276211		
		172RG	691, 172RG0001		
		FP172	FP172-0001 Thru FP172-0003		
		F172D Thru F172P	F172-0001 Thru F17202254		
		FR172E Thru FR172K	FR17200001 Thru FR17200675		

Continue Airworthin Program Inspectio Number	d ness Inspection n Compliance Title	Model	Effectivity	Initial	Repeat
28-30-01	Quick Drain Valves -	175 Thru 175C	619, 28700A,	1000 Hrs	1000 Hrs
Fu	Fuel Reservoir and Fuel Tank Inspection	177 Thru 177B	55001 Thru 17557119 661, 17700001	or 2 Yrs	2Yrs
		177RG	177RG0001		
		F177RG	F177RG0001 Thru		
		180 Thru 180K	604, 614, 624, 645,		
		182 Thru 182R	613, 631, 634, 675,		
		R182	R1820001 Thru		
		A182J Thru A182N F182P Thru F182O	A182-0001 Thru A182-0148 F18200001 Thru E18200169		
		F182RG	FR18200001 Thru		
		185 Thru A185F	632, 652, 185-0001 Thru 18504448		
		188 Thru T188C	653, 678T, 188-0001		
		A-A188B	A-A1880001 Thru A-A188034		
		190/195/195A/ 195B	7004 Thru 16183		
28-30-02	Strainer Drain Control	150 Thru 150M	617, 628, 644, 649, 17001 Thru 15079405	5 Yrs Control	5 Yrs Control
		A150K Thru	A1500001	5 Yrs	5 Yrs
		A150M	Thru A1500734	Spring	Spring
		A-150L	A-1501001 Thru		
			A-1501039		
		A-A150L	A-A1500001 Thru		
		EIFOE The	A-A1500009		
		FISUE INFU	F 150-0001 Thru E 15001429		
		FA150K Thru	FA15000428		
		FA150M	FA1500336		
		152	15279406		
		A152	Thru 15286033 681, A1500433, A1520735		
		F152	F15201429 Thru F15201980		
		FA152	FA1520337 Thru FA1520425		

Continued Airworthine Program Inspection Number	ess Inspection Compliance Title	Model	Effectivity	Initial	Repeat
28-30-02	Strainer Drain Control Inspection	172 Thru 172Q	609, 1800 Thru 27169 610, 612, 615, 622, 625, 630, 638, 639, 2800 Thru 17276211	5 Yrs Control 5 Yrs Spring	5 Yrs Control 5 Yrs Spring
		P172D	P17257120 Thru P17257188	opg	-F3
		R172E Thru 172K	680, R172-0001 Thru R1723454		
		172RG	691, 177RG0001 Thru 177RG1191		
		FP172	FP172-0001 Thru FP172-0003		
		F172D Thru F172P	F172-0001 Thru F17202254		
		FR172E Thru FR172K	FR17200001 Thru FR17200675		
		175 Thru 175C	619, 28700A, 55001 Thru 17557119		
		177 Thru 177B	661, 17700001 Thru 17702752		
		177RG F177RG	177RG0001 F177RG0001 Thru F177RG0177 Thru 177RG1366		
		180 Thru 180K	604, 614, 624, 645, 30000 Thru 18053203		
		182 Thru 182R	613, 631, 634, 675, 33000 Thru 18268586		
		R182 A182J Thru	R1820001 A182-0001 Thru		
		A182N	A182-0148 Thru 18202039		
		F182P Thru F182Q	F18200001 Thru F18200169		
		F182RG	FR18200001 Thru FR18200070		
		185 Thru A185F	632, 652, 185-0001 Thru 18504448		
		188 Thru 188C	653, 6781, 188-0001 Thru T18803974T		
		A-A188B	A-A1880001 Thru A-A188034		

Continued Airworthine Program Inspection Number	ess	Inspection Compliance Title	Mode!	Effectivity	Initial	Repeat
Number 28-40-01	Fuel	Title Gaging System ection	Model 120 140 140 140 150 Thru 150M A150K Thru A150K Thru A-A150L F150F Thru F150F Thru F150M FA150M F152 F172 F172D F172 F172 F172D F172D F172 F172D F172D	Effectivity 8003 Thru 15075 8001 Thru 15075 15200 Thru 15724 617, 628, 644, 649, 17001 Thru 15079405 A1500001 Thru A1500734 A-1501003 A-A1500001 Thru A-1501039 A-A1500001 Thru A-1500009 F150-0001 Thru F15001428 FA1500001 Thru FA1500336 15279406 Thru 15286033 681, A1500433, A1520735 Thru A1521049 F15201429 Thru F15201980 FA1520337 Thru F15201980 FA1520337 Thru FA1520425 609, 1800 Thru 27169 610, 612, 615, 622, 625, 630, 638, 639, 2800 Thru 17276211 P17257188 680, R172-0001 Thru R1723454 172RG0001 Thru 177RG1366 FP172-0001 Thru F172022	Initial 1000 Hrs or 3 Yrs	Repeat 1000 Hrs or 3 Yrs
			F172P FR172E Thru FR172K 175 Thru 175C 177 Thru 177B	F17202254 FR17200001 Thru FR17200675 619, 28700A, 55001 Thru 17557119 661, 17700001 Thru 17702752		

Continued Airworthine Program Inspection	ess	Inspection Compliance	Madal	F ffeetister		Descet
Number		The	wodei	Enectivity	initial	нереат
28-40-01	Fuel Inspe	Gaging System	177RG	177RG0001 Thru 177RG1366	1000 Hrs or 3 Yrs	1000 Hrs or 3 Yrs
			F177RG	F177RG0001 Thru F177RG0177		
			180 Thru 180K	604, 614, 624, 645, 30000 Thru 18053203		
			182 Thru 182R	613, 631, 634, 675, 33000 Thru 18268586		
			R182	R1820001		
			A182J Thru	A182-0001 Thru		
			A182N	A182-0148		
			F182P Thru	F18200001 Thru		
			F182Q	F18200169		
			105 Thus 41055	Thru 18202039		
			185 Inru A185F	632, 652, 185-0001 Thru 18504448		
			188 Thru 188C	653, 678T, 188-0001		
				Thru T18803974T		
			A-A188B	A-A1880001 Thru A-A188034		
29-10-01	Hydr	aulic Hose	120	8003 Thru 15075	1000 Hrs	1000 Hrs
	Repla	acement and	140	8001 Thru 15075	or 3 Yrs	or 3 Yrs
	Hydr	aulic Component	140A	15200 Thru 15724		
	Repla	acement	150 Thru 150M	617, 628, 644, 649, 17001 Thru 15079405		
			A150K Thru	A1500001		
			A150M	Thru A1500734		
			A-150L	A-1501001 Ihru		
			A-A150	A-1501039 A-A1500001 Thru		
			AAIOOL	A-A1500001 mild		
			F150F Thru	F150-0001 Thru		
			F150M	F15001428		
			FA150K Thru	FA1500001 Thru		
			FA150M	FA1500336		
			152	15279406		
				Thru 15286033		
			A152	681, A1500433,		
				A1520735 Thru		
			F150	A1521049		
			F152	152/9406 Thru		
			FA152	10200000 EA1500227 Their		
			TATUZ	FA1520327 Thru FA1520425		

Continued Airworthin	t less				
Program	Inspection				
Inspectior Number	n Compliance Title	Model	Effectivity	Initial	Repeat
29-10-01	Hydraulic Hose Replacement and Hydraulic Component Replacement	170 Thru 170B 172 Thru 172Q	609, 1800 Thru 27169 610, 612, 615, 622, 625, 630, 638, 639, 2800 Thru 17276211	1000 Hrs or 3 Yrs	1000 Hrs or 3 Yrs
		P172D	P17257120		
		R172E Thru 172	Thru P17257188 < 680, R172-0001		
		172RG	691, 172RG0001 Thru 172RG1191		
		FP172	FP172-0001 Thru FP172-0003		
		F172D Thru F172P	F172-0001 Thru F17202254		
		FR172E Thru	FR17200001 Thru		
		FR172K	FR17200675		
		175 Thru 175C	619, 28700A, 55001		
		177 Thru 177B	661, 17700001 Thru 17702752		
		177RG	177RG0001 Thru 177RG1366		
		F177RG	F177RG0001 Thru F177RG0177		
		180 Thru 180K	604, 614, 624, 645, 30000 Thru 18053203		
		182 Thru 182R	613, 631, 634, 675, 33000 Thru 18268586		
		R182	R1820001 Thru 18202039		
		A182J Thru	A182-0001 Thru		
		A182N	A182-0148		
		F182P Thru	F18200001 Thru		
		F182Q	F18200169		
		F182RG	FR18200001 Thru FR18200070		
		185 Thru A185F	632, 652, 185-0001 Thru 18504448		
		188 Thru 188C	653, 678T, 188-0001 Thru T18803974T		
		A-A188B	A-A1880001 Thru A-A188034		
		190/195/195A/ 195B	7004 Thru 16183		

Continued Airworthine	Continued Airworthiness					
Program Inspection Number		Inspection Compliance Title	Model	Effectivity	Initial	Repeat
32-10-00	Main Pivot	Landing Gear Inspection	172RG R182 F182RG	680, 172RG0001 Thru 172RG1191 R18200001 Thru R182202039 FR18200001 Thru FR18200070	1000 Hrs	1000 Hrs
32-10-01	Main Outb Inspe	Landing Spring oard Support ection	150 Thru 150M A150K Thru A150M A-150L A-A150L F150F Thru F150M FA150K Thru FA150K Thru FA150M 180 Thru 180K 185 Thru A185F	617, 628, 644, 649, 17001 Thru 15079405 A150001 Thru A1500734 A-1501001 Thru A-1501039 A-A1500001 Thru A-A1500009 F150-0001 Thru F150-0067 FA1500001 Thru FA1500336 604, 614, 624, 645, 30000 Thru 18053203 632, 652, 185-0001 Thru 18504448	1000 Hrs or 3 Yrs	1000 Hrs or 3 Yrs
32-10-02	Main Outb Gear Inspe	Landing Spring oard Support Support Forging ection	172P Thru 172Q F172 Thru F172N	17257162 Thru 17267584 F172-0560 Thru F17201749	1000 Hrs	1000 Hrs
32-30-01	Main Actua Bolts	Landing Gear ator Mounting	172RG R182 F182RG	691, 172RG0001 Thru 172RG1191 R18200001 Thru R18202039 FR18200001 Thru FR18200070	1000 Hrs	None
32-40-00	Main Gear	Landing Axle Inspection	120 140 140A 170A	8003 Thru 15075 8001 Thru 15075 15200 Thru 15724 18730 Thru 20266	2000 Hrs	1000 Hrs

Continued Airworthin Program Inspection	d less Inspection n Compliance				
Number	Title	Model	Effectivity	Initial	Repeat
52-10-00	Upper Door Hinge Inspection	177 Thru 177B 177RG F177RG	661, 17700001 17702752 177RG0001 Thru 177RG1366 F177RG0001 Thru F177RG0177	1000 Hrs	500 Hrs
53-10-00	Landing Gear Bulkhead Inspection	150 Thru 150M FA150K Thru FA150M	617, 628, 644, 649, 17001 Thru 15079405 FA1500001 Thru FA1500336	500 Hrs	300 Hrs
53-10-01	Fuselage Strut Attach Area	180 Thru 180K 182 Thru 182R R182 A182J Thru A182N F182P Thru F182Q F182RG 185 Thru A185F	604, 614, 624, 645, 30000 Thru 18053203 613, 631, 634, 675 33000 Thru 1826586 R1820001 Thru 18202039 182-0001 Thru A182-0148 F18200001 Thru F18200169 FR18200001 Thru FR18200070 632, 652, 185-0001 Thru 18504448	12,000 Hrs	s 2000 Hrs
53-10-02	Fuselage Wing Root Rib Inspection	177 Thru 177B 177RG F177RG	661, 17700001 Thru 17702752 177RG0001 Thru 177RG1366 F177RG0001 Thru F177RG0177	12,000 Hrs	s 2000 Hrs
53-10-03	Brake Cylinder Mounting Bracket and Stiffener Inspection	172RG	691, 172RG0001 Thru 172RG1191	1000 Hrs	500 Hrs

Continued Airworthin	l ess					
Program		Inspection				
Number	•	Title	Model	Effectivity	Initial	Repeat
53-20-00	Bagg Corr	gage Compartment osion	182 Thru 182R R182 A182J Thru A182N F182P Thru F182Q F182RG	613, 631, 634, 675, 33000 Thru 18268586 R1820001 Thru 18202039 A182-0001 Thru A182-0148 F18200001 Thru F18200169 FR18200001 Thru FR18200070	1000 Hrs or 2 Yrs	1000 Hrs or 2 Yrs
53-40-00	Verti	cal Fin Attachments	180 Thru 180K 182 Thru 182R R182 A182J Thru A182N F182P Thru F182Q F182RG 185 Thru A185F	604, 614, 624, 645, 30000 Thru 18053203 613, 631, 634, 675, 33000 Thru 18268586 R1820001 Thru 18202039 A182-0001 Thru A182-0148 F18200001 Thru F18200169 FR18200001 Thru FR18200070 632, 652, 185-0001 Thru 18504448	12,000 Hrs	s 6000 Hrs
53-40-01	Wing Attac	g Fuselage chment	120 140 177 Thru 177B	8003 Thru 15075 8001 Thru 15075 661, 17700001 Thru 17702752	12,000 Hrs	s 2000 Hrs
53-40-02	Inspe Spar	ect Carry Thru Corrosion	177 Thru 177B 177R	661, 17700001 Thru 17702752 177RG0001 Thru 177RG1366	1000 Hrs or 2 Yrs	1000 Hrs 2 Yrs
55-10-01	Horiz Attac	zontal Stabilizer chment Inspection	150 Thru 150M A150K Thru A150L A-A150L F150F Thru F150M	617, 628, 644, 649, 17001 Thru 15072099 A1500001 Thru A1500238 A-A1500001 Thru A-A1500009 F150-0001 Thru F15001428	1000 Hrs	1000 Hrs

Continued Airworthin Program Inspection Number	ess Inspection Compliance Title	Model	Effectivity	Initial	Repeat
55-10-01	Horizontal Stabilizer Attachment Inspection	FA150K Thru FA150M	FA1500001 Thru FA1500336	1000 Hrs	1000 Hrs
55-10-02	Horizontal Stabilizer Attachment Brackets Inspection	177 Thru 177B	661, 17700001 Thru 17701370	1000 Hrs	1000 Hrs
55-10-03	Horizontal and Vertical Stabilizers Attachment Nuts Inspection	172M F172M	17256493 Thru 17267584 F17200905 Thru F17201514	3000 Hrs	1500 Hrs
55-20-00	Elevator Trim Tab Inspection	188 Thru T188C A-A188B	653, 678T, 188-0001 Thru T18803974T A-A1880001 Thru A-A188034	1000 Hrs	1000 Hrs
55-20-01	Stabilizer Balance Weight Bracket Inspection	177 Thru 177B 177RG F177RG	661, 17700001 Thru 17702752 177RG0001 Thru 177RG1366 F177RG0001 Thru F177RG0177	5000 Hrs	200 Hrs
55-20-02	Elevator Outboard Hinge Bolt and Nutplate Inspection	152 152A F152	15279406 Thru 15285833 A1520735 Thru Thru A1521025 F15201429 Thru F15201943	1000 Hrs	500 Hrs
55-30-00	Vertical Fin Attach Bracket Inspection	150 Thru 150M 150M 152 A152	15077789 Thru 15079405 A1500654 Thru A1500734 15279406 Thru 15282617 681, A1500433, A1520735 Thru A1520833	1000 Hrs	500 Hrs

Continued Airworthine Program Inspection Number	ess Inspection Compliance Title	Model	Effectivity	Initial	Repeat
55-30-00	Vertical Fin Attach Bracket Inspection	F152 FA152	F15201429 Thru F15201980 FA1520337 Thru	1000 Hrs	500 Hrs
55-30-01	Vertical Fin Rear Spar Inspection	180K 185F 188B T188C A-A188B	FA1520425 18052385 Thru 18053147 18502311 Thru 18504070 18801375 Thru 18803725 T18803307T Thru T18803725T A-A1880001 Thru A-A188034	500 Hrs	500 Hrs
55-30-02	Vertical Fin Attach Bracket Nutplate Inspection	150 Thru 150M 150K Thru 150M 152 A152 F152 FA152	617, 628, 644, 649, 17001 Thru 15079405 A1500001 Thru A1500734 15279406 Thru 15286033 A1520735 Thru A1521049 F15201429 Thru F15201980 FA1520337 Thru FA1520425	500 Hrs	200 Hrs
55-30-03	Vertical Stabilizer Attachment Inspection	182E Thru 182R A182J Thru A182N F182P Thru F182Q F182RG	634, 675, 18253599 Thru 18268586 A182-0001 Thru A182-0148 F18200001 Thru F18200169 FR18200001 Thru FR18200070	1000 Hrs	1000 Hrs
55-40-00	Rudder Stop Clip Inspection	182Q F182Q	18266591 Thru 18267715 F18200026 Thru F18200169	1000 Hrs or 3 Yrs	1000 Hrs 3 Yrs

SECTION II - LISTING OF CONTINUED AIRWORTHINESS PROGRAM INSPECTIONS

Continued Airworthin Program Inspectior Number	d less Inspection n Compliance Title	Model	Effectivity	Initial	Repeat
57-10-00	Aileron Balance Weight	150 Thru 150M	617, 628, 644, 649,	3000 Hrs	3000 Hrs
	Inspection		17001 Thru 15079405		
		150K Thru 150M	A1500001 Thru A1500734		
		A-150L	A-1501001 Thru A-1501039		
		A-A150L	A-A1500001 Thru		
		E150E Thru	E150-0001 Thru		
		F150M	F15001428		
		FA150K Thru	FA1500001 Thru		
		FA150M	FA1500336		
		172 Thru 172Q	610, 612, 615, 622,		
			625, 630, 638,		
			2800 Thru 17276211		
		R172E Thru	R1722000 Thru		
		R172K	R1723454		
		172RG	691, 172RG0001		
		D.470D	Thru 172RG1191		
		P1/2D	P1/25/20 Inru		
			P1/25/188		
		FP1/2	FP172-0001 Inru		
		E172D Thru	FF 172-0003 F172-0001 Thru		
		F172D 1110	F17202254		
		FB172E Thru	FR17200001 Thru		
		FR172K	FR17200675		
			Thru P17257188		
		180 Thru 180K	604, 614, 624, 645,		
			30000 Thru 18053203		
		182 Thru 182R	613, 631, 634, 675,		
			33000 Thru 18268586		
		R182	R1820001		
			Thru 18202039		
		A182J Ihru	A182-0001 Ihru		
		ATOZN	A 182-0148		
			F1020001 11110		
		F102Q	F10200103		
		1 10200	FR1820007		
		185 Thru A185F	632, 652, 185-0001		
			Thru 18504448		

Continued Airworthing Program Inspection Number	ess	Inspection Compliance Title	Model	Effectivity	Initial	Repeat
	145					•
57-10-01	Wing Hole	spar Spray Boom	188B	18801178 Thru 18802057	1000 Hrs	1000 Hrs
57-10-02	Wing	g Strut and End	120	8003 Thru 15075	12,000 Hrs	2000 Hrs
	Fittin	ng	140	8001 Thru 15075		
			140A	15200 Thru 15724		
			150 Thru 150M	617, 628, 644, 649,		
				17001 Thru 15079405		
			A150K Thru	A1500001		
			A150M	Thru A1500734		
			A-150L	A-1501001 Thru		
				A-1501039		
			A-A150L	A-A1500001 Thru		
				A-A1500009		
			F150F Thru	F150-0001 Thru		
			F150M	F15001428		
			FA150K Thru	FA1500001 Thru		
			FA150M	FA1500336		
			152	15279406		
				Thru 15286033		
			A152	681, A1500433,		
				A1520735		
				Thru A1521049		
			170 Thru 170B	609, 1800 Thru 27169		
			172 Thru 172Q	610, 612, 615, 622,		
				625, 630, 638, 639,		
				2800 Thru 17276211		
			P172D	P17257120		
				Thru P17257188		
			R172E Thru 172K	680, R172-0001		
				Thru R1723454		
			172RG	691, 172RG0001		
				Thru 17RG1191		
			FP172	FP172-0001 Thru		
				FP172-0003		
			F172D Thru	F172-0001 Thru		
			F172P	F17202254		
			FR172E Thru	FR17200001 Thru		
			FR172K	FR17200675		
			175 Thru 175C	619, 28700A, 55001		
				Thru 17557119		
			180 Thru 180K	604, 614, 624, 645,		
				30000 Thru 18053203		
			182 Thru 182R	613, 631, 634, 675,		
				33000 Thru 18268586		

Continued Airworthiness							
Program		Inspection					
Number		Title	Model	Effectivity	Initial	Repeat	
57-10-02	Wing Fittin	g Strut and End	R182 A182J Thru A182N F182P Thru F182Q F182RG 185 Thru A185F	R1820001 Thru 18202039 A182-0001 Thru A182-0148 F18200001 Thru F18200169 FR18200001 Thru FR18200070 632, 652, 185-0001 Thru 18504448	12,000 Hrs	2000 Hrs	
57-10-03	Wing Fittin	g Fuselage Attach ng Inspection	120 140 140A 150 Thru 150M	8003 Thru 15075 8001 Thru 15075 15200 Thru 15724 617, 628, 644, 649, 17001 Thru 15079405	12,000 Hrs	3000 Hrs	
57-10-03	Wing	g Fuselage Attach ng Inspection	A150K Thru A150M A-150L A-A150L F150F Thru F150M FA150K Thru FA150M 152 A152 A152 170 Thru 170B 172 Thru 172Q P172D R172E Thru 172K 172RG FP172	A1500001 Thru A1500734 A-1501001 Thru A-1501039 A-A1500009 F150-0001 Thru F15001428 FA1500001 Thru F15001428 FA1500001 Thru FA1500336 15279406 Thru 15286033 681, A1500433, A1520735 Thru A1521049 609, 1800 Thru 27169 610, 612, 615, 622, 625, 630, 638, 639, 2800 Thru 17276211 P17257120 Thru P17257188 680, R172-0001 Thru R1723454 691, 172RG0001 Thru 17RG1191 FP172-0001 Thru FP172-0001 Thru	12,000 Hrs	3000 Hrs	

Continued Airworthin Program Inspection Number	ess Inspection Compliance Title	Model	Effectivity	Initial	Repeat
57-10-03	Wing Fuselage Attach Fitting Inspection	F172D Thru F172P FR172E Thru FR172K 175 Thru 175C	F172-0001 Thru F17202254 FR17200001 Thru FR17200675 619, 28700A, 55001 Thru 17557119	12,000 Hrs	3000 Hrs
		180 Thru 180K	604, 614, 624, 645, 30000 Thru 18053203		
		162 Mru 162R	33000 Thru 18268586		
		R182	R1820001 Thru 18202039		
		A182J Thru A182N F182P Thru F182Q F182RG	A182-0001 Thru A182-0148 F18200001 Thru F18200169 FR18200001 Thru EB182000070		
		185 Thru A185F	632, 652, 185-0001 Thru 18504448		
57-10-05	Wing Fuselage Attach Fitting Rear Spar Center	177 Thru 177B	661, 17700001 Thru 17702752	12,000 Hrs	3000 Hrs
		177RG	177RG0001		
		F177RG	F177RG0001 Thru F177RG0177		
		190/195/195A /195B	7004 Thru 16183		
57-20-00	Wing Spar Inspection	188 Thru 188A	188-0001 Thru 18800832	1000 Hrs	1000 Hrs
		A-A188B	A-A1880001 Thru A-A18803		
57-50-00	Flap Inspection	120 140 140A 150 Thru 150M A150K Thru	8003 Thru 15075 8001 Thru 15075 15200 Thru 15724 617, 628, 644, 649, 17001 Thru 15079405 A1500001	1000 Hrs 🔇	500 Hrs
		A150M F150F Thru F150M	Thru A1500734 F150-0001 Thru F15001428		
Continued Airworthine Program Inspection	ss Inspection Compliance				
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Number	Title	Model	Effectivity	Initial	Repeat
57-50-00	Flap Inspection	FA150K Thru FA150M 152	FA1500001 Thru FA1500336 15279406 Thru 15286033	1000 Hrs	500 Hrs
		A152	681, A1500433, A1520735 Thru A1521049		
		F152	F15201429 Thru F15201980		
		FA152	FA1520337 Thru FA1520425		
		170 Thru 170B 172 Thru 172Q	609, 1800 Thru 27169 610, 612, 615, 622, 625, 630, 638, 639, 2800 Thru 17276211		
		P172D	P17257120 Thru P17257188		
		R172E Thru 172K	680, R172-0001 Thru R1723454		
		172RG	691, 177RG0001 Thru 177RG1366		
		FP172	FP172-0001 Thru FP172-0003		
		F172D Thru	F172-0001 Thru		
		FR172E Thru	FR17200001 Thru		
		FR172K	FR17200675		
		175 Thru 175C	619, 28700A, 55001 Thru 17557119		
		177 Thru 177B	661, 17700001 Thru 17702752		
		177RG	177RG0001 Thru 177RG1366		
		F177RG	F177RG0001 Thru F177RG0177		
	,	180 Thru 180K	604, 614, 624, 645, 30000 Thru 18053203		
		182 Thru 182R	613, 631, 634, 675, 33000 Thru 18268586		
		R182	R1820001 Thru 18202039		
		F182P Thru F182Q	F18200001 Thru F18200169		

Continued Airworthiness						
Program		Inspection				
Inspection		Compliance				
Number		litle	Model	Effectivity	Initial	Repeat
57-50-00	Flap	Inspection	F182RG	FR18200001 Thru FR18200070	1000 Hrs	500 Hrs
			A182J Thru	A182-0001 Thru		
			A182N	A182-0148		
			185 Thru A185F	632, 652, 185-0001 Thru 18504448		
			188 Thru 188C	653, 678T, 188-0001 Thru T18803974T		
			A-A188B	A-A1880001 Thru A-A188034		
			190/195/195A/ 195B	7004 Thru 16183		
57-50-01	Engi	ne Mount Weld	120	8003 Thru 15075	1000 Hrs	500 Hrs
	Joint	Inspection	140	8001 Thru 15075		
			1 40A	15200 Thru 15724		
			150 Thru 150M	617, 628, 644, 649,		
				17001 Thru 15079405		
			A150K Thru	A1500001		
			A150M	Thru A1500734		
			F150F Thru	F150-0001 Thru		
			F150M	F15001428		
			FA150K Thru	FA1500001 Thru		
			FA150M	FA1500336		
			152	15279406		
				Thru 15286033		
			A152	681, A1500433,		
				A1520735 Thru		
			F 1 F 0	A1521049		
			F152	F15201429 Inru		
			EA150	F 15201960		
			FA152	FA1520337 Infu		
			170 Thru 170P	FA1020420		
			172 Thru 1720	610 612 615 622		
				625 630 638 639		
				2800 Thru 17276211		
			P172D	P17257120		
				Thru P17257188		
			R172E Thru 172K	680, R172-0001		
				Thru R1723454		
			172RG	691, 177RG0001		
				Thru 177RG1366		
			FP172	FP172-0001 Thru		
				FP172-0003		

Continued Airworthiness					
Program	Inspection				
Inspection	n Compliance	Madal		lus istin l	Demost
Number	nue	Woder	Ellectivity	initial	кереат
57-50-01	Engine Mount Weld	F172D Thru	F172-0001 Thru	1000 Hrs	500 Hrs
	Joint Inspection	F172P	F17202254		
		FR172E Thru	FR17200001 Thru		
		FR172K	FR17200675		
		175 Thru 175C	619, 28700A, 55001		
			Thru 17557119		
		180 Thru 180K	604, 614, 624, 645,		
			30000 Thru 18053203		
		182 Thru 182R	613, 631, 634, 675,		
		D.(00	33000 I hru 18268586		
		R182	R1820001		
			Thru 18202039		
		F 182P Inru	F18200001 Inru		
		F182Q	F18200169		
		F 162RG	FR 18200001 Infu		
		A182 Thru	A182.0001 Thru		
		A 1820 MINU	A182-0149		
		185 Thru A185E	632 652 185-0001		
			Thru 18504448		
		188 Thru 188C	653, 678T, 188-0001		
			Thru T18803974T		
		A-A188B	A-A1880001 Thru		
			A-A188034		
57-60-00	Aileron Inspection	120	8003 Thru 15075	1000 Hrs	500 Hrs
		140	8001 Thru 15075		
		140A	15200 Thru 15724		
		150 Thru 150M	617, 628, 644, 649,		
			17001 Thru 15079405		
		A150K Thru	A1500001		
		A150M	Thru A1500734		
		F150F Thru	F150-0001 Thru		
		F150M	F15001428		
		FA150K Thru	FA1500001 Thru		
		FA150M	FA1500336		
		152	152/9406 The 4500000		
		A150	INTU 15286033		
		A102	001, A1000433, A1500725 Three		
			A1520/35 INTU		
		F152	F1521049		
		102	F15201980		
		FA152	FA1520337 Thru		
			FA1520425		

Continued Airworthine	ess					
Program		Inspection				
Inspection Number	1	Compliance Title	Model	Effectivity	Initial	Repeat
57-50-01	Ailer	on Inspection	170 Thru 170B 172 Thru 172	609, 1800 Thru 27169 610, 612, 615, 622, 625, 630, 638, 639, 2800 Thru 17276211	1000 Hrs	500 Hrs
			P172D	P17257120 Thru P17257188		
			R172E Thru 172K	680, R172-0001 Thru R1723454		
			172RG	691, 177RG0001 Thru 177RG1366		
			FP172	FP172-0001 Thru FP172-0003		
			F172D Thru F172P	F172-0001 Thru F17202254		
			FR172E Thru	FR17200001 Thru		
			FR172K	FR17200675		
			175 Inru 1750	519, 28700A, 55001 Thru 17557119		
			177 Thru 177B	661, 17700001 Thru 17702752		
			177RG	177RG0001 Thru 177RG1366		
			F177RG	F177RG0001 Thru F177RG0177		
			180 Thru 180K	604, 614, 624, 645, 30000 Thru 18053203		
			182 Thru 182R	613, 631, 634, 675, 33000 Thru 18268586		
			R182	R1820001 Thru 18202039		
			F182P Thru	F18200001 Thru		
			F182Q	F18200169		
			F 182RG	FR1820007		
			A182J Thru	A182-0001 Thru		
			A182N	A182-0148		
			185 Inru A185F	632, 652, 185-0001 Thru 18504448		
			188 Thru 188C	653, 678T, 188-0001 Thru T18803974T		
			A-A188B	A-A1880001 Thru A-A188034		
			190/195/195A/ 195B	7004 Thru 16183		

Continued Airworthin Program Inspectior Number	i less Inspection Compliance Title	Model	Effectivity	Initial	Repeat
71-20-00	Engine Mount Welded	120 140	8003 Thru 15075 8001 Thru 15075	10,000 Hrs	5000 Hrs
		140A 150 Thru 150M	15200 Thru 15724 617, 628, 644, 649, 17001 Thru 15070405		
		A150K Thru	A1500001 Thru A1500734		
		E150E Thru	E150-0001 Thru		
		F150M	F15001428		
		FA150K Thru	FA1500001 Thru		
		FA150M	FA1500336		
		152	15279406		
			Thru 15286033		
		A152	681, A1500433,		
			A1520735 Thru		
		5450	A1521049		
		F152	F15201429 Inru		
		FA152	F 15201960 F 1520337 Thru		
		17102	FA1520425		
		170 Thru 170B	609. 1800 Thru 27169		
		172 Thru 172	610, 612, 615, 622,		
			625, 630, 638, 639,		
			2800 Thru 17276211		
		P172D	P17257120		
			Thru P17257188		
		R172E Thru	680, R172-0001		
		172K	I Nru K1/23454		
		17260	Thru 177RG1366		
		FP172	FP172-0001 Thru		
			FP172-0003		
		F172D Thru	F172-0001 Thru		
		F172P	F17202254		
		FR172E Thru	FR17200001 Thru		
		FR172K	FR17200675		
		175 Thru 175C	619, 28700A, 55001		
		177 The 1770	INTU 1/557119		
		177 INFU 1778	17702752		
		177RG	177RG0001 Thru		
		177NG	177RG1366		
		F177RG	F177RG0001 Thru		
			F177RG0177		

Continued Airworthiness					
Program	Inspection				
Number	Title	Model	Effectivity	Initial	Repeat
71-20-00	Engine Mount Welded Joint Inspection	180 Thru 180K	604, 614, 624, 645, 30000 Thru 18053203	10,000 Hrs	5000 Hrs
		182 Thru 182R	613, 631, 634, 675, 33000 Thru 18268586		
		R182	R1820001 Thru 18202039		
		F182P Thru	F18200001 Thru		
		F182Q	F18200169		
		F182RG	FR18200001 Thru FR18200070		
		A182J Thru	A182-0001 Thru		
		A182N	A182-0148		
		185 Thru A185F	632, 652, 185-0001 Thru 18504448		
		188 Thru 188C	653, 678T, 188-0001 Thru T18803974T		
		A-A188B	A-A1880001 Thru A-A188034		
		190/195/195A/ 195B	7004 Thru 16183		
71-20-01	Inspect Engine Mount For Corrosion	180 Thru 180K	604, 614, 624, 645, 30000 Thru 18053203	10,000 Hrs or 5 Yrs o	5000 Hrs or 3 Yrs
		182 Thru 182R	613, 631, 634, 675, 33000 Thru 18268586		
		R182	R1820001 Thru 18202039		
		A182J Thru	A182-0001 Thru		
		A182N	A182-0148		
		F182P Thru	F18200001 Thru		
		F182Q	F18200169		
		F182RG	FR18200001 Thru FR18200070		
71-60-00	Cowl Flap Hinge Pin Inspection	180 Thru 180K	604, 614, 624, 645, 30000 Thru 18053203	1000 Hrs ⁻ or 2 Yrs o	1000 Hrs or 2 Yrs
		182 Thru 182R	613, 631, 634, 675, 33000 Thru 18268586		
		R182	R1820001 Thru 18202039		
	·	A182J Thru	A182-0001 Thru		
		A182N	A182-0148		
		F182P Thru	F18200001 Thru		
		F182Q	F1820069		

Contine Airwort Prograe Inspect Numbe	ued hiness m Inspection tion Compliance r Title	Model	Effectivity	Initial	Repeat
71-60-0	0 Cowl Flap Hinge Pin Inspection	F182RG	FR18200001 Thru FR18200070	1000 Hrs or 2 Yrs	1000 Hrs or 2 Yrs
		185 Thru A185F	632, 652, 185-0001 Thru 18504448		
79-10-0	0 Oil Cooler Inspection	152	15279406 Thru 15286033	1000 Hrs or 3 Yrs	1000 Hrs or 3 Yrs
		A152	A1520735 Thru A1521049		
		F152	F15201429 Thru F15201980		
		FA152	FA1520337 Ihru FA1520425		
		172 1110 1720	625, 630, 638, 639, 2800 Thru 17276211		
		P172D	P17257120 Thru P17257188		
		172RG	691, 172RG0001 Thru 172RG1191 EP172-0001 Thru		
		F172D Thru	FP172-0003 F172-0001 Thru		
		F172P	F17202254		
		FR172E Thru	FR17200001 Thru		
		FR172K	FR17200675		
		1// Inru 1//B	661, 17700001 Thru 17702752		
		177RG	177RG0001 Thru 177RG1366		
		182 Thru 182D	F177RG0001 Inru F177RG0177		
		R182	33000 Thru 18268586 B1820001		
		F182P Thru	Thru 18202039 F18200001 Thru		
		F182Q F182RG	F18200169 FR18200001 Thru FR18200070		
		A182J Thru	A182-0148		
		A182N	A182-0001 Thru		

TITLE Control Wheel Inspection

MODEL

EFFECTIVITY

150 Thru 150M	617, 628, 644, 649, 17001 Thru 150794	105 INSPE	CTION COMPLIANCE
A150K Thru A150M	A1500001 Thru A1500734		
A-150L	A-1501001 Thru A-1501039	INITIAL	1000 HRS
A-A150L	A-A1500001 Thru A-A1500009		
F150F Thru F150M	F150-0001 Thru F15001428	REPEAT	1000 HRS
FA150K Thru FA150M	FA1500001 Thru FA1500336		Thereafter
172 Thru 172Q	610, 612, 615, 622, 625, 630, 638,		
	639, 2800 Thru 17276211		
P172D	P17257120 Thru P17257188		
FP172	FP172-0001 Thru FP172-0003		
F172D Thru F172P	F172-0001 Thru F17202254		
FR172E Thru FR172K	FR17200001 Thru FR17200675		
175 Thru 175C	619, 28700A, 55001 Thru 17557119		
180 Thru 180K	604, 614, 624, 645, 30000 Thru 180532	203	
182 Thru 182R	613, 631, 634, 675, 33000 Thru 182685	586	
R182	R1820001 Thru 18202039		
A182J Thru A182N	A182-0001 Thru A182-0148		
F182P Thru F182Q	F18200001 Thru F18200169		
F182RG	FR18200001 Thru FR18200070		
188 Thru T188C	653, 678T, 188-0001 Thru T18803974T		
A-A188B	A-A1880001 Thru A-A188034		
•			

PURPOSE

To inspect control wheel for cracks.

INSPECTION INSTRUCTIONS

Visual inspect control wheel for cracks.

ACCESS/LOCATION

Control Column

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Replace control wheel.

COMMENTS

Ref: SL64-8

DETECTABLE CRACK SIZE

TITLE Aileron End Bearing Inspection

MODEL

EFFECTIVITY

 177 Thru 177B
 661, 17700001 Thru 17702752
 INSPECTION COMPLIANCE

 177RG
 177RG0001 Thru 177RG1366
 INITIAL 1000 HRS

 F177RG
 F177RG0001 Thru F177RG0177
 INITIAL 1000 HRS

 188 Thru T188C
 653, 678T, 188-0001 Thru T18803974T
 or 3 YRS

 A-A188B
 A-A1880001 Thru A-A188034
 REPEAT 1000 HRS

 or 3 YRS
 or 3 YRS

PURPOSE

To visual inspect aileron end for corrosion.

INSPECTION INSTRUCTIONS

Disconnect rod end at the aileron, inspect rod end for corrosion.

ACCESS/LOCATION

Aileron

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

If corrosion is found, replace rod end.

COMMENTS

Ref: SE67-60

DETECTABLE CRACK SIZE

Thereafter

TITLE Aileron Cable Inspection

MODEL

EFFECTIVITY

 188 Thru T188A
 653, 678T, 188-0001 Thru
 INSPECTION COMPLIANCE

 1880707
 1880707

 A-A188B
 A-A1880001 Thru A-A188034
 INITIAL 500 HRS

REPEAT 500 HRS Thereafter

PURPOSE

To provide for aileron cable replacement every 500 hours.

INSPECTION INSTRUCTIONS

Remove and replace aileron cables every 500 hours in accordance with Service Manual.

ACCESS/LOCATION

Fuselage/Wing

INSPECTION PROCEDURE

None

REPAIR/MODIFICATION

Replacement

COMMENTS

Ref: SE73-12

DETECTABLE CRACK SIZE

TITLE Rudder Cable Inspection

MODEL	EFFECTIVITY		
152	15279406 Thru 15286033	INSPECTION C	OMPLIANCE
A152	681, A1500433,		
	A1520735 Thru A1521049	INITIAL	1000 HRS
F152	F15201429 Thru F15201980		
FA152	FA1520337 Thru FA1520425	REPEAT	1000 HRS Thereafter

PURPOSE

To visually inspect rudder cables for broken strands.

INSPECTION INSTRUCTIONS

Visually inspect rudder cables for broken strands 7.5 inches to 9.00 inches from turnbuckle end.

ACCESS/LOCATION

Tailcone

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Replace in accordance with applicable Service Manual.

COMMENTS

DETECTABLE CRACK SIZE

TITLE

Hose - Fuel Line Inspection

MODEL

EFFECTIVITY

150 Thru 150M A150K Thru A150M	617, 628, 644, 649, 17001 Thru 15079405 A1500001 Thru A1500734	INSPEC	TION COMPLIANCE
A-150L	A-1501001 Thru A-1501039	INITIAL	5 YRS
A-A150L	A-A1500001 Thru A-A1500009		
F150F Thru F150M	F150-0001 Thru F15001428	REPEAT	2 YRS
FA150K Thru FA150M	FA1500001 Thru FA1500336		Thereafter
152	15279406 Thru 15286033		
A152	681, A1500433, A1520735 Thru A1521049		
F152	F15201429 Thru F15201980		
FA152	FA1520337 Thru FA1520425		
170 Thru 170B	609, 1800 Thru 27169		
172 Thru 172Q	610, 612, 615, 622, 625, 630, 638,		
	639, 2800 Thru 17276211		
P172D	P17257120 Thru P17257188		
R172E Thru 172K	680, R172-0001 Thru R1723454		
172RG	691, 172RG0001 Thru 172RG1191		
FP172	FP172-0001 Thru FP172-0003		
F172D Thru F172P	F172-0001 Thru F17202254		
FR172E Thru FR172K	FR17200001 Thru FR17200675		
175 Thru 175C	619, 28700A, 55001 Thru 17557119		
177 Thru 177B	661, 17700001 Thru 17702752		
177RG	177RG0001 Thru 177RG1366		
F177RG	F177RG0001 Thru F177RG0177		
180 Thru 180K	604, 614, 624, 645, 30000 Thru 18053203		
182 Thru 182R	613, 631, 634, 675, 33000 Thru 18268586		
R182	R1820001 Thru 18202039		
A182J Thru A182N	A182-0001 Thru A182-0148		
F182P Thru F182Q	F18200001 Thru F18200169		
F182RG	FR18200001 Thru FR18200070		
185 Thru A185F	632, 652, 185-0001 Thru 18504448		
188 Ihru 188C	653, 678T, 188-0001 Thru T18803974T		
A-A188B	A-A1880001 Thru A-A188034		
190/195/195A/195B	7004 Thru 16183		

PURPOSE

To ensure hose integrity.

INSPECTION INSTRUCTIONS

- 1. Inside cabin, open up the headliner upholstery just forward of rear door post.
- 2. Inside wing, remove access cover next to cabin on bottom side.
- 3. Check hose on each side for cracks, weeping and deterioration.
- 4. Replace hose as necessary. When replacing hoses, use new clamps.

Note: Refer to applicable Service Manual for location of hoses.

ACCESS/LOCATION

Cabin Area

DETECTABLE CRACK SIZE

N/A

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Replace hose as required.

COMMENTS

TITLE Hose Inspection - Fuel Vent

MODEL

EFFECTIVITY

180 Thru 180K	604, 614, 624, 645, 30000 Thru 18053203	INSPECTION COMPLIANCE
182 Thru 182D	613, 631, 634, 675, 33000 Thru R18202041	
A182J Thru A182N	A182-0001 Thru A182-0148	INITIAL 5 YRS
F182P Thru F182Q	F18200001 Thru F18200169	
F182RG	FR18200001 Thru FR18200070	REPEAT 2 YRS
185 Thru A185F	632, 652, 185-0001 Thru 18504448	Thereafter

PURPOSE

To ensure hose integrity.

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INSPECTION INSTRUCTIONS

- 1. Check hose on each side for cracks, weeping and deterioration.
- 2. Replace hose as necessary. When replacing hoses, use new clamps.

ACCESS/LOCATION

Access plate just outboard of fuel tank on bottom side wing.

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Replace hose as required.

COMMENTS

DETECTABLE CRACK SIZE



Fuel Vent Hose Inspection Figure 3-1 (Sheet 1)

TITLE Fuel Vent Line Inspection

MODEL

EFFECTIVITY

177 Thru 177B	661, 17701974 Thru 17702123	INSPECTION COMPLIANCE
177RG	177RG0433 Thru 177RG0592	
F177RG	F177RG0001 Thru F177RG0177	INITIAL 1000 HRS

REPEAT 1000 HRS Thereafter

PURPOSE

To inspect fuel vent line for possible kinks or line collapse.

INSPECTION INSTRUCTIONS

- 1. Break safety wire at drain fitting and remove wing to fuselage fairing.
- 2. Visually inspect for kinks or collapsed vent line.

ACCESS/LOCATION

Wing to Fuselage

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Replace fuel vent line in accordance with applicable Service Manual.

COMMENTS

Ref: SE74-14

DETECTABLE CRACK SIZE



Fuel Vent Line Inspection Figure 3-2 (Sheet 1)

TITLE

Quick Drain Valves - Fuel Reservoir and Fuel Tanks Inspection

MODEL

EFFECTIVITY

150 Thru 150M A150K Thru A150M	617, 628, 644, 649, 17001 Thru 150794 A1500001 Thru A1500734	105 INSP	ECTION COMPLIANCE
A-150L	A-1501001 Thru A-1501039	INITIAL	1000 HRS
A-A150L	A-A1500001 Thru A-A1500009		or 2 YRS
F150F Thru F150M	F150-0001 Thru F15001428	REPEAT	1000 HRS
FA150K Thru FA150M	FA1500001 Thru FA1500336		or 2 YRS
152	15279406 Thru 15286033		Thereafter
A152	681, A1500433, A1520735 Thru A15210	049	
170 Thru 170B	609, 1800 Thru 27169		
172 Thru 172Q	610, 612, 615, 622, 625, 630, 638,		
	639, 2800 Thru 17276211		
172RG	691, 172RG0001 Thru 172RG1191		
FP172	FP172-0001 Thru FP172-0003		
F172D Thru F172P	F172-0001 Thru F17202254		
FR172E Thru FR172K	FR17200001 Thru FR17200675		
175 Thru 175C	619, 28700A, 55001 Thru 17557119		
177 Thru 177B	661, 17700001 Thru 17702752		
177RG	177RG0001 Thru 177RG1366		
F177RG	F177RG0001 Thru F177RG0177		
180 Thru 180K	604, 614, 624, 645, 30000 Thru 180532	203	
182 Thru 182R	613, 631, 634, 675, 33000 Thru 182685	686	
R182	R1820001 Thru 18202039		
A182J Thru A182N	A182-0001 Thru A182-0148		
F182P Thru F182Q	F18200001 Thru F18200169		
F182RG	FR18200001 Thru FR18200070		
185 Thru A185F	632, 652, 185-0001Thru 18504448		
188 Thru T188C	653, 678T, 188-0001 Thru T18803974T		
A-A188B	A-A1880001 Thru A-A188034		
190/195/195A/195B	7004 Thru 16183		

PURPOSE

To check for possible corrosion.

INSPECTION INSTRUCTIONS

- 1. Remove fuel reservoir/fuel tank quick drain valve and inspect for corrosion. If corrosion exists the valve must be replaced.
- 2. Inspect fuel reservoir for internal corrosion.

ACCESS/LOCATION

DETECTABLE CRACK SIZE

N/A

Under side of wing.

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Replace as required.

COMMENTS

Ref: SE84-8



FUEL DRAIN VALVE

Quick Drain Valves - Fuel Reservoir Fuel Tank Figure 3-3 (Sheet 1)

TITLE

Strainer Drain Control Inspection

MODEL

EFFECTIVITY

150 Thru 150M	617, 628, 644, 649, 17001 Thru 150794	05 INSP	INSPECTION COMPLIANCE	
A150K Thru A150M	A1500001 Thru A1500734			
A-150L	A-1501001 Thru A-1501039	INITIAL	5 YRS - Control	
A-A150L	A-A1500001 Thru A-A1500009		5 YRS - Spring	
F150F Thru F150M	F150-0001 Thru F15001428	REPEAT	5 YRS - Control	
FA150K Thru FA150M	FA1500001 Thru FA1500336		5 YRS - Spring	
152	15279406 Thru 15286033		Thereafter	
A152	681, A1500433, A1520735 Thru A15210	49		
F152	F15201429 Thru F15201980			
FA152	FA1520337 Thru FA1520425			
170 Thru 170B	609, 1800 Thru 27169			
172 Thru 172Q	610, 612, 615, 622, 625, 630, 638,			
	639, 2800 Thru 17276211			
P172D	P17257120 Thru P17257188			
R172E Thru 172K	680, R172-0001 Thru R1723454			
172RG	691, 172RG0001 Thru 172RG1191			
FP172	FP172-0001 Thru FP172-0003			
F172D Thru F172P	F172-0001 Thru F17202254			
FR172E Thru FR172K	FR17200001 Thru FR17200675			
175 Thru 175C	619, 28700A, 55001 Thru 17557119			
177 Thru 177B	661, 17700001 Thru 17702752			
177RG	177RG0001 Thru 177RG1366			
F177RG	F177RG0001 Thru F177RG0177			
180 Thru 180K	604, 614, 624, 645, 30000 Thru 180532	03		
182 Thru 182R	613, 631, 634, 675, 33000 Thru 182685	86		
R182	R1820001 Thru 18202039			
A182J Thru A182N	A182-0001 Thru A182-0148			
F182P Thru F182Q	F18200001 Thru F18200169			
F182RG	FR18200001 Thru FR18200070			
185 Thru A185F	632, 652, 185-0001 Thru 18504448			
188 Thru T188C	653, 678T, 188-0001 Thru T18803974T			
A-A188B	A-A1880001 Thru A-A188034			

PURPOSE

If fuel strainer drain valve does not close fully when control is released, it can cause continuous loss of fuel.

INSPECTION INSTRUCTIONS

- 1. Ensure that control is rigged properly, refer to applicable Service Manual.
- 2. Accomplish preflight check; pull control to drain strainer, then manually push control forward after actuating. Visually check that valve is closed.

ACCESS/LOCATION

DETECTABLE CRACK SIZE

Lower instrument panel/lower firewall.

INSPECTION PROCEDURE

Visual REPAIR/MODIFICATION

Replace control/return spring.

COMMENTS

TITLE

Fuel Gaging System Inspection

MODEL

100

EFFECTIVITY

8003 Thru 15075

120
140
140A
150 Thru 150M
A150K Thru A150M
A-150L
A-A150L
F150F Thru F150M
FA150K Thru FA150M
152
A152
F152
FA152
170 Thru 170B
172 Thru 172Q
P172D
R172E Thru 172K
172RG
FP172
F172D Thru F172P
FR172E Thru FR172K
175 Thru 175C
177 Thru 177B
177RG
F177RG
180 Thru 180K
182 Thru 182R
R182
A182J Thru A182N
F182P Thru F182Q
F182RG
185 Thru A185F
188 Thru T188C
A-A188B

8001 Thru 15075 15200 Thru 15724 617, 628, 644, 649, 17001 Thru 15079405 A150001 Thru A1500734 A-1501001 Thru A-1501039 A-A1500001 Thru A-A1500009 F150-0001 Thru F15001428 FA1500001 Thru FA1500336 15279406 Thru 15286033 681, A1500433, A1520735 Thru A1521049 F15201429 Thru F15201980 FA1520337 Thru FA1520425 609, 1800 Thru 27169 610, 612, 615, 622, 625, 630, 638, 639, 2800 Thru 17276211 P17257120 Thru P17257188 680, R172-0001 Thru R1723454 691, 172RG0001 Thru 172RG1191 FP172-0001 Thru FP172-0003 F172-0001 Thru F17202254 FR17200001 Thru FR17200675 619, 28700A, 55001 Thru 17557119 661,17700001 Thru 17702752 177RG0001 Thru 177RG1366 F177RG0001 Thru F177RG0177 604, 614, 624, 645, 30000 Thru 18053203 613, 631, 634, 675, 33000 Thru 18268586 R1820001 Thru 18202039 A182-0001 Thru A182-0148 F18200001 Thru F18200169 FR18200001 Thru FR18200070 632, 652, 185-0001 Thru 18504448 653, 678T, 188-0001 Thru T18803974T A-A1880001 Thru A-A188034

INSPECTION COMPLIANCE

INITIAL 1000 HRS or 3 YRS REPEAT 1000 HRS or 3 YRS Thereafter

PURPOSE

To ensure that fuel gage is reading properly.

INSPECTION INSTRUCTIONS

- 1. Defuel airplane.
- 2. Turn master switch On; ensure that fuel gage reads "0" empty when tank has only unusable fuel remaining.

ACCESS/LOCATION

DETECTABLE CRACK SIZE

Wing tank filler port.

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Adjust/replace transmitter and/or gage as required.

COMMENTS

TITLE

Hydraulic Hose Replacement and Hydraulic Component Replacement

MODEL

EFFECTIVITY

120 140 140A 150 Thru 150M A150K Thru A150M A-150L A-A150L F150F Thru F150M FA150K Thru FA150M 152 A152 F152 FA152 170 Thru 170B 172 Thru 172Q P172D R172E Thru 172K 172RG FP172 F172D Thru F172P FR172E Thru FR172K 175 Thru 175C 177 Thru 177B 177RG **F177RG** 180 Thru 180K 182 Thru 182R R182 A182J Thru A182N F182P Thru F182Q F182RG 185 Thru A185F 188 Thru T188C A-A188B 190/195/195A/195B

8003 Thru 15075 INSPECTION COMPLIANCE 8001 Thru 15075 15200 Thru 15724 INITIAL 1000 HRS 617, 628, 644, 649, 17001 Thru 15079405 A150001 Thru A1500734 REPEAT 1000 HRS A-1501001 Thru A-1501039 A-A1500001 Thru A-A1500009 F150-0001 Thru F15001428 FA1500001 Thru FA1500336 15279406 Thru 15286033 681, A1500433, A1520735 Thru A1521049 F15201429 Thru F15201980 FA1520337 Thru FA1520425 609, 1800 Thru 27169 610, 612, 615, 622, 625, 630, 638, 639, 2800 Thru 17276211 P17257120 Thru P17257188 680, R172-0001 Thru R1723454 691, 172RG0001 Thru 172RG1191 FP172-0001 Thru FP172-0003 F172-0001 Thru F17202254 FR17200001 Thru FR17200675 619, 28700A, 55001 Thru 17557119 661, 17700001 Thru 17702752 177RG0001 Thru 177RG1366 F177RG0001 Thru F177RG0177 604, 614, 624, 645, 30000 Thru 18053203 613, 631, 634, 675, 33000 Thru 18268586 R1820001 Thru 18202039 A182-0001 Thru A182-0148 F18200001 Thru F18200169 FR18200001 Thru FR18200070 632, 652, 185-0001 Thru 18504448 653, 678T, 188-0001 Thru T18803974T A-A1880001 Thru A-A188034 7004 Thru 16183

PURPOSE

To ensure that the hydraulic hose and components installed on the airplane are inspected at the proper interval.

INSPECTION INSTRUCTIONS

Refer to SNL85-54. (See excerpts under COMMENTS of this CAP.)

ACCESS/LOCATION

DETECTABLE CRACK SIZE

or 3 YRS

or 3 YRS

Thereafter

As required.

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

COMMENTS

Ref: SNL85-54.

The "on condition" interval applies to -- *

- * Brake and landing gear hydraulic system hoses of the later synthetic material (S2178-XXX part numbers).
- * Landing gear hydraulic system power packs used on 1979 and later models (9881124-XX).
- * All other components except earlier power packs and hoses of rubber material.

Landing gear power packs used prior to 1979 models still require overhaul every 5 years to replace rubber components. All brake and landing gear hydraulic system hoses used prior to the S2178-XXX part numbers still require replacement every 5 years to replace the rubber hose material.

NOTE

Although part number identification is the positive method to determine hose material, the synthetic hydraulic system hoses generally have a smooth, somewhat shiny surface texture and are generally a blue or reddish color. Rubber hoses are generally black and have a dull, rougher surface texture.

The new overhaul/replacement requirements will be incorporated in future revisions to the applicable airplane Parts Catalogs and Service Manuals.

The revised requirements in no way preclude the importance of accomplishing thorough hydraulic system inspections, as detailed in the applicable airplane Service Manual. Routine inspections are of the utmost importance to ensure continued airworthiness, durability and reliability.

TITLE Main Landing Gear Pivot Inspection

MODEL EFFECTIVITY

172RG	172RG0001 Thru 172RG1191	INSPECT	TION COMPLIANCE
R182	R18200477 Thru R182202041		
F182RG	FR18200001 Thru FR18200070	INITIAL	1000 HRS

REPEAT 1000 HRS Thereafter

PURPOSE

To inspect main landing pivot assembly for cracks in spline area.

INSPECTION INSTRUCTIONS

Fluorescent penetrant inspect main landing gear pivot assembly in accordance with SEB90-1.

ACCESS/LOCATION

Fuselage

INSPECTION PROCEDURE

Fluorescent Penetrant

REPAIR/MODIFICATION

Replace landing gear pivot in accordance with SEB90-1 Revision 2.

COMMENTS

Ref: SEB90-1

DETECTABLE CRACK SIZE

TITLE Main Landing Springs Outboard Support Inspection

MODEL

EFFECTIVITY

150 Thru 150M	617, 628, 644, 649, 17001 Thru 15079405	INSPECTION	COMPLIANCE
A150K Thru A150M	A150001 Thru A1500734		
A-150L	A-1501001 Thru A-1501039	INITIAL	1000 HRS
A-A150L	A-A1500001 Thru A-A1500009		or 3 Yrs
F150F Thru F150M	F150-0001 Thru F15001428	REPEAT	1000 HRS
FA150K Thru FA150M	FA1500001 Thru FA1500336		or 3 Yrs
180 Thru 180K	604, 614, 624, 645, 30000 Thru 18053203		Thereafter
185 Thru A185F	632, 652, 185-0001 Thru 18504448		

PURPOSE

To inspect main landing gear spring outboard support for corrosion.

INSPECTION INSTRUCTIONS

- 1. Remove main landing gear springs from airplane in accordance with Service Manual.
- 2. Visually inspect outboard spring support for corrosion.

ACCESS/LOCATION

DETECTABLE CRACK SIZE

Fuselage

N/A

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

If corrosion is found, contact Cessna Aircraft Company and describe condition.

COMMENTS



Main Landing Springs Outboard Support Inspection Figure 3-4 (Sheet 1)

3-26 **32-10-01** Apr 3/95

TITLE Main Landing Spring Outboard Gear Support Forging Inspection

MODEL EFFECTIVITY

172K Thru 172N	17257162 Thru 17267584	INSPECTION COMPLIANCE
F172 Thru F172N	F172-0560 Thru F17201749	

INITIAL 1000 HRS

REPEAT 1000 HRS Thereafter

PURPOSE

To visually inspect main landing gear spring outboard support forging for cracking in web.

INSPECTION INSTRUCTIONS

- 1. Remove main landing in accordance with Service Manual.
- 2. Visually inspect main landing gear spring outboard support forging for cracks in web.

ACCESS/LOCATION

Fuselage under floor panels.

DETECTABLE CRACK SIZE

N/A

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

If cracks are found, contact Cessna Aircraft Company and describe condition.

COMMENTS



Main Landing Spring Outboard Gear Support Forging Inspection Figure 3-5 (Sheet 1)

3-28 **32-10-02** Apr 3/95

TITLE Main Landing Gear Actuator Mounting Bolts

MODEL EFFECTIVITY 172RG 691, 172RG0001 Thru 172RG1191 INSPECTIO R182 R18200001 Thru R18202039 INITIAL 10 F182RG FR18200001 Thru FR18200070 INITIAL 10

INSPECTION COMPLIANCE

INITIAL 1000 HRS

REPEAT None

PURPOSE

Bolts without safety wire provisions have been found loose during routine inspections.

INSPECTION INSTRUCTIONS

Remove actuator bolts without safety wire previsions and replace with AN174H22A or AN175H22A bolts and secure with safety wire.

ACCESS/LOCATION

Remove center access panel on top of main landing gear bulkheads.

DETECTABLE CRACK SIZE

N/A

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

COMMENTS

Ref: SNL86-49





REPLACE EXISTING BOLT WITH AN174H22A OR AN175H22 BOLT

> Main Landing Gear Actuator Mounting Bolts Figure 3-6 (Sheet 1)

TITLE Main Landing Gear Axle Inspection

MODEL	EFFECTIVITY		
120	8003 Thru 15075	INSPECTION C	OMPLIANCE
140	8001 Thru 15075		
140A	15200 Thru 15724	INITIAL	2000 HRS
170A	18730 Thru 20266		
		REPEAT	1000 HRS

PURPOSE

To ensure main landing gear structural integrity.

INSPECTION INSTRUCTIONS

- 1. Remove main landing gear axle from main gear in accordance with applicable Service Manual.
- 2. Magnetic Particle inspect main landing gear axle for cracks.
- 3. If no cracks are found, reinstall axle in main landing gear.

ACCESS/LOCATION

Landing Gear

INSPECTION PROCEDURE

Magnetic Particle

REPAIR/MODIFICATION

Replace axle in accordance with applicable Service Manual.

COMMENTS

DETECTABLE CRACK SIZE

TITLE Upper Door Hinge Inspection

MODEL EFFECTIVITY

177 Thru 177B	661, 17700001 Thru 17702752	INSPECTION C	OMPLIANCE
177RG	177RG0001 Thru 177RG1366		
F177RG	F177RG0001 Thru F177RG0177	INITIAL	1000 HRS

REPEAT 500 HRS Thereafter

PURPOSE

To ensure that door hinge operates properly.

INSPECTION INSTRUCTIONS

Visually inspect door hinge for elongation and proper lubrication.

ACCESS/LOCATION

Fuselage

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Replace door hinge if required in accordance with applicable Service Manual. Lubricate door hinge with medium weight - LPS2.

COMMENTS

DETECTABLE CRACK SIZE

TITLE Landing Gear Bulkhead Inspection

MODEL EFFECTIVITY

150 Thru 150M	617, 628, 644, 649, 17001 Thru 15079405	INSPECTION COMPLIANCE
FA150K Thru FA150M	FA1500001 Thru FA1500336	

INITIAL 500 HRS

REPEAT 300 HRS Thereafter

PURPOSE

To check for possible damage (cracks) in landing gear outboard support bracket and/or bulkhead assemblies.

INSPECTION INSTRUCTIONS

Inspect for any cracks in the landing outboard support bracket and/or bulkhead assemblies.

ACCESS/LOCATION

DETECTABLE CRACK SIZE

N/A

Remove upholstery and floorboards to gain access.

INSPECTION PROCEDURE

Visually inspect for cracks in landing outboard support bracket and/or bulkhead assemblies.

REPAIR/MODIFICATION

If any cracks are found, install SK150-6.

COMMENTS

TITLE Fuselage Strut Attach Area

MODEL

EFFECTIVITY

180 Thru 180K	604, 614, 624, 645, 30000 Thru 18053203	INSPECT	ION COMPLIANCE
182 Thru 182R	613, 631, 634, 675, 33000 Thru 18268586	6	
R182	R1820001 Thru 18202039	INITIAL	12,000 HRS
A182J Thru A182N	182-0001 Thru A182-0148		
F182P Thru F182Q	F18200001 Thru F18200169	REPEAT	2,000 HRS
F182RG	FR18200001 Thru FR18200070		Thereafter
185 Thru A185F	632, 652, 185-0001 Thru 18504448		

PURPOSE

To inspect fuselage strut attach area web for cracks.

INSPECTION INSTRUCTIONS

Visually inspect for cracks in web on the outboard end of bulkhead, inboard of strut attach area.

ACCESS/LOCATION

Inside fuselage, under floor panel.

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

If cracks are found, contact Cessna Aircraft Company and describe condition.

COMMENTS

DETECTABLE CRACK SIZE


Fuselage Strut Attach Area Figure 3-7 (Sheet 1)

3-38 **53-10-01** Apr 3/95

TITLE Fuselage Wing Root Rib Inspection

MODEL

EFFECTIVITY

177 Thru 177B	661, 17700001 Thru 17702752	INSPECTION COMPLIANCE
177RG	177RG0001 Thru 177RG1366	
F177RG	F177RG0001 Thru F177RG0177	INITIAL 12,000 HRS

REPEAT 2,000 HRS Thereafter

PURPOSE

To inspect fuselage wing root rib bend radius for cracks.

INSPECTION INSTRUCTIONS

Visually inspect wing root rib bend radius for cracks.

ACCESS/LOCATION

Inside fuselage under headliner.

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

If crack is found, contact Cessna Aircraft Company and describe condition.

COMMENTS

DETECTABLE CRACK SIZE



Fuselage Wing Root Inspection Figure 3-8 (Sheet 1)

Brake Cylinder Mounting Bracket and Stiffener Inspection

MODEL

TITLE

EFFECTIVITY

172RG

691, 172RG0001 Thru 172RG1191

INSPECTION COMPLIANCE

INITIAL 1000 HRS

REPEAT 500 HRS Thereafter

PURPOSE

To inspect brake cylinder mounting bracket and stiffener for cracks.

INSPECTION INSTRUCTIONS

- 1. Remove rudder pedal bar shields in accordance with Service Manual.
- 2. Visually inspect mounting bracket and stiffener for cracks.

ACCESS/LOCATION

Below right rudder pedal on pilot's side.

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

- 1. Remove brake cylinder in accordance with Service Manual.
- 2. Replace damaged bracket and stiffener. Refer to Parts Catalog for replacement parts.
- 3. Install brake cylinder in accordance with Service Manual.
- 4. Reinstall rudder pedal bar shields in accordance with Service Manual.

COMMENTS

N/A

DETECTABLE CRACK SIZE

TITLE Baggage Compartment Corrosion

MODEL

EFFECTIVITY

 182 Thru 182R
 613, 631, 634, 675, 33000 Thru 18268586
 INSPECTION COMPLIANCE

 R182
 R1820001 Thru 18202039
 INITIAL
 1000 HRS

 A182J Thru A182N
 A182-0001 Thru A182-0148
 INITIAL
 1000 HRS

 F182P Thru F182Q
 F18200001 Thru F18200169
 or 2 YRS

 F182RG
 FR18200001 Thru FR18200070
 REPEAT
 1000 HRS

 or 2 YRS
 Or 2 YRS
 Or 2 YRS

PURPOSE

To inspect baggage compartment panels for corrosion.

INSPECTION INSTRUCTIONS

- 1. Remove carpet as required to gain access to panel.
- 2. Visually inspect panel and structure for corrosion.

ACCESS/LOCATION

Tailcone

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Remove corrosion using standard corrosion removal procedures.

COMMENTS

Ref: SL65-1

DETECTABLE CRACK SIZE

Thereafter

TITLE Vertical Fin Attachments

MODEL

EFFECTIVITY

180 Thru 180K 182 Thru 182R R182 A182J Thru A182N F182P Thru F182Q F182RG 185 Thru A185F **purpose** 604, 614, 624, 645, 30000 Thru 18053203 613, 631, 634, 675, 33000 Thru 18268586 R1820001 Thru 18202039 A182-0001 Thru A182-0148 F18200001 Thru F18200169 FR18200001 Thru FR18200070 632, 652, 185-0001 Thru 18504448

INSPECTION COMPLIANCE

INITIAL 12,000 HRS

REPEAT 6000 HRS Thereafter

Structural Testing

INSPECTION INSTRUCTIONS

- 1. Visual inspect forward and aft bulkheads in area of vertical fin attachment.
- 2. Dye Penetrant inspect rudder cable cutouts in forward and aft bulkheads.
- 3. Dye Penetrant inspect the upper corners of aft bulkheads at horizontal stabilizer attachments.

ACCESS/LOCATION

Remove stinger and fin to horizontal tail fairings.

INSPECTION PROCEDURE

Visual and Dye Penetrant

REPAIR/MODIFICATION

If any damage is found, contact Cessna Aircraft Company.

COMMENTS

Ref: SE72-29

DETECTABLE CRACK SIZE

0.06 inch



Vertical Fin Attachments Figure 3-9 (Sheet 1)



Vertical Fin Attachments Figure 3-9 (Sheet 2)



Figure 3-9 (Sheet 3)



Vertical Fin Attachments Figure 3-9 (Sheet 4)

E Wing Fuselage Attachment

MODEL

EFFECTIVITY

120 140 177 Thru 177B 8003 Thru 15075 8001 Thru 15075 661, 17700001 Thru 17702752 **INSPECTION COMPLIANCE**

INITIAL 12,000 HRS

REPEAT 2000 HRS Thereafter

PURPOSE

To inspect wing attach fitting for damage.

INSPECTION INSTRUCTIONS

- 1. Eddy Current inspect wing attach fittings and adjacent supporting structure.
- 2. Replace cracked wing attach fitting and bolts.
- 3. Eddy Current inspect per instructions is section 4-1, Basic Information, pages x and xi.

ACCESS/LOCATION

Remove wing-to-fuselage fairings.

INSPECTION PROCEDURE

Eddy Current

REPAIR/MODIFICATION

If cracks are found, contact Cessna Aircraft Company and describe condition.

COMMENTS

Temporary Revision Number 1 July 15, 1998 3-51

53-40-01

DETECTABLE CRACK SIZE

N/A

TITLE



Fuselage Wing Attachment Figure 3-10 (Sheet 1)

TITLE Inspect Carry Thru Spar Corrosion

MODEL

EFFECTIVITY

177 Thru 177B 177R 661, 17700001 Thru 17702752 177RG0001 Thru 177RG1366 **INSPECTION COMPLIANCE**

INITIAL 1000 HRS or 2 YRS REPEAT 1000 HRS or 2 YRS Thereafter

PURPOSE

To inspect Carry thru Spar for corrosion.

INSPECTION INSTRUCTIONS

- 1. Remove headliner as required to gain access to carry thru spar.
- 2. Visually inspect carry thru spar and structure for corrosion.

ACCESS/LOCATION

Nose compartment

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Remove corrosion using standard corrosion removal procedures.

COMMENTS

DETECTABLE CRACK SIZE

TITLE Horizontal Stabilizer Attachment Inspection

EFFECTIVITY

150 Thru 150M	617, 628, 644, 649, 17001 Thru 15072099	INSPECTION COMPLIANCE
A150K Thru A150L	A1500001 Thru A1500238	
A-150L	A-1501001 Thru A-1501039	INITIAL 1000 HRS
A-A150L	A-A1500001 Thru A-A1500009	
F150F Thru F150M	F150-0001 Thru F15001428	REPEAT 1000 HRS
FA150K Thru FA150M	FA1500001 Thru FA1500336	Thereafter

PURPOSE

To inspect horizontal stabilizer forward attach stiffener for cracks.

INSPECTION INSTRUCTIONS

Visually inspect forward attach stiffener for cracks.

ACCESS/LOCATION

Horizontal stabilizer/tailcone

DETECTABLE CRACK SIZE

Minor Cracks: No more than one per side and less than 0.75 inch in length. Extensive Cracking: More than the above

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

- 1. Stop drill minor cracks and install a doubler in accordance with SK150-33.
- 2. For extensive cracking, repair in accordance with SE71-23.

COMMENTS

TITLE Horizontal Stabilizer Attachment Brackets Inspection

MODEL

EFFECTIVITY

177 Thru 177B

661, 17700001 Thru 17701370

INSPECTION COMPLIANCE

INITIAL 1000 HRS

REPEAT 1000 HRS Thereafter

PURPOSE

To inspect attachment brackets for cracks.

INSPECTION INSTRUCTIONS

Eddy Current inspect per instructions is section 4-1, Basic Information, pages x and xi. If attachment brackets are cracked, refer to SK177-11A for repair.

ACCESS/LOCATION

Horizontal Stabilizer

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Refer to SK177-11A.

COMMENTS

Ref: SE70-32

DETECTABLE CRACK SIZE

TITLE Horizontal and Vertical Stabilizers Attachment Nuts Inspection

MODEL EFFECTIVITY

172M F172M	17256493 Thru 17267584 F17200905 Thru F17201514	INSPECTION C	OMPLIANCE
		INITIAL	3000 HRS

INITIAL 3000 HRS

REPEAT 1500 HRS Thereafter

PURPOSE

To Horizontal and Vertical Stabilizers Attachment Nuts for cracks.

INSPECTION INSTRUCTIONS

Visually inspect Horizontal and Vertical Stabilizers attachment nuts for cracks.

ACCESS/LOCATION

Tailcone

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Replace existing nuts in accordance with Service Manual.

COMMENTS

DETECTABLE CRACK SIZE

Elevator Trim Tab Horn Inspection

EFFECTIVITY

MODEL

188 Thru T188C 653, 678T, 188-0001 Thru T18803974T INSPECTION COMPLIANCE A-A188B A-A1880001 Thru A-A188034

INITIAL 1000 HRS

REPEAT 1000 HRS Thereafter

PURPOSE

To check elevator trim tab horn for looseness and cracks around skin where trim tab horn attaches to trim tab.

INSPECTION INSTRUCTIONS

- 1. Check for looseness of elevator trim tab horn.
- 2. Inspect for crack appearing in skin where trim tab horn attaches to the trim tab.

ACCESS/LOCATION

Elevator Trim Tab

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

If either condition exists replace trim tab.

COMMENTS

Ref: SE80-57

DETECTABLE CRACK SIZE

N/A

TITLE

TITLE Stabilizer Balance Weight Bracket Inspection

MODEL

EFFECTIVITY

177 Thru 177B	661, 17700001 Thru 17702752	INSPECTION COMPLIANCE
177RG	177RG0001 Thru 177RG1366	
F177RG	F177RG0001 Thru F177RG0177	INITIAL 5000 HRS

REPEAT 200 HRS Thereafter

PURPOSE

To inspect stabilizer balance weight brackets for cracks.

INSPECTION INSTRUCTIONS

Visually inspect balance weight bracket for cracks in accordance with SEB89-1.

ACCESS/LOCATION

Tailcone

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Replace balance weight bracket. (Refer to SEB89-1.)

COMMENTS

Ref: SEB89-1

DETECTABLE CRACK SIZE

Elevator Outboard Hinge Bolt and Nutplate Inspection

MODEL

EFFECTIVITY

152	15279406 Thru 15285833
A152	A1520735 Thru A1521025
F152	F15201429 Thru F15201943

INSPECTION COMPLIANCE

INITIAL 1000 HRS

REPEAT 500 HRS Thereafter

PURPOSE

To visually inspect outboard hinge bolt and nutplate for security.

NOTE: If elevator has been replaced in accordance with SE83-26, this inspection procedures is not applicable.

INSPECTION INSTRUCTIONS

Inspect outboard hinge bolt for security of installation.

ACCESS/LOCATION

Elevator outboard hinge

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

If bolt was found to be loose, remove existing bolt and discard, install a longer AN3 Bolt, AN310-3 Nut and Cotter Pin.

COMMENTS

DETECTABLE CRACK SIZE

N/A

TITLE

TITLE Vertical Fin Attach Bracket Inspection

EFFECTIVITY

150 Thru 150M	15077789 Thru 15079405	INSPEC	TION COMPLIANCE
150M	A1500654 Thru A1500734		
152	15279406 Thru 15282617	INITIAL	1000 HRS
A152	681, A1500433, A1520735 Thru A1520833		
F152	F15201429 Thru F15201980	REPEAT	500 HRS
FA152	FA1520337 Thru FA1520425		Thereafter

PURPOSE

To visually inspect vertical fin attach fitting for cracks.

INSPECTION INSTRUCTIONS

Visual inspect vertical fin attach fitting for cracks. If cracks are present, attach fitting must be replaced.

ACCESS/LOCATION

DETECTABLE CRACK SIZE

Tailcone

N/A

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

- 1. Remove vertical fin in accordance with applicable Service Manual.
- 2. Remove damaged attach fitting and discard.
- 3. Position replacement attach fitting in place on vertical fin and back drill holes in new attach fitting to 0.193 inch diameter matching existing holes in vertical fin.
- 4. Deburr holes and secure attach fitting to vertical fin with existing bolts. Torque bolts to 40 inchpounds.
- 5. Install vertical fin in accordance with applicable Service Manual.

COMMENTS

Ref: SE78-62



Vertical Fin Attach Fitting Figure 3-11 (Sheet 1)

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TITLE Vertical Fin Rear Spar Inspection

180K	18052385 Thru 18053147	INSPECTION COMPLIANCE
185F	18502311 Thru 18504070	
188B	18801375 Thru 18803725	INITIAL 500 HRS
T188C	T18803307T Thru T18803725T	-
A-A188B	A-A1880001 Thru A-A188034	REPEAT 500 HRS
		Thereafter

PURPOSE

MODEL

To inspect vertical fin rear spar for cracks.

INSPECTION INSTRUCTIONS

Visually inspect vertical fin rear spar flange for cracks.

ACCESS/LOCATION

Inspect through elevator torque tube cut outs in tailcone stinger.

EFFECTIVITY

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Replace vertical fin rear spar assembly in accordance with applicable Service Kit. (Refer to SE80-85.)

COMMENTS

Ref: SE80-85

DETECTABLE CRACK SIZE



Vertical Fin Rear Spar Inspection Figure 3-12 (Sheet 1)

TITLE Vertical Fin Attach Bracket Nutplate Inspection

MODEL

EFFECTIVITY

150 Thru 150M	617, 628, 644, 649, 17001 Thru 15079405	INSPECT	ION COMPLIANCE
A150K Thru A150M	A1500001 Thru A1500734		
A-150L	A-1501001 Thru A-1501039	INITIAL	500 HRS
A-A150L	A-A1500001 Thru A-A1500009		
F150F Thru F150M	F150-0001 Thru F15001428	REPEAT	200 HRS
FA150K Thru FA150M	FA1500001 Thru FA1500336		Thereafter
152	15279406 Thru 15286033		
A152A	A1520735 Thru A1521049		
F152	F15201429 Thru F15201980		
FA152	FA1520337 Thru FA1520425		

PURPOSE

To inspect vertical fin attach bracket nutplates for cracks.

INSPECTION INSTRUCTIONS

- 1. Position rudder to full left. Inspect nutplates on upper and lower flange of attach bracket for cracks.
- 2. Position rudder to full right. Inspect nutplates on upper and lower flange of attach bracket for cracks.

ACCESS/LOCATION

DETECTABLE CRACK SIZE

N/A

Stabilizer rear spar.

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

- 1. Remove rudder, elevator, and vertical fin in accordance with applicable Service Manual.
- 2. Remove bolts securing stabilizer rear spar to fuselage tailcone.
- 3. Remove rivets securing attach bracket to stabilizer spar and attach bracket.
- 4. Remove damaged nutplates and discard.

NOTE: MS21042L4 or MS2104N4 Nut will be used in place of the nutplate.

- 6. Install attach bracket on stabilizer spar, securing with rivets.
- 7. Install rudder, elevator, and vertical fin in accordance with applicable Service Manual.

COMMENTS

Ref: SE79-49

TITLE Vertical Stabilizer Attachment Inspection

EFFECTIVITY

MODEL

182E Thru 182R	634, 675, 18253599 Thru 18268586	INSPECTION C	OMPLIANCE
A182J Thru A182N 5182P Thru 51820	A182-0001 Thru A182-0148	INITIAL	1000 HRS
-182RG	FR18200001 Thru FR18200070	REPEAT	1000 HRS Thereafter

PURPOSE

Structural Testing

INSPECTION INSTRUCTIONS

- 1. Visually and dye penetrant inspect forward and aft vertical fin attachment for cracks, particularly the shaded areas. refer to Figure.
- 2. Check attach bolts for looseness.
 - (a) If bolts are loose remove bolts and dye penetrant inspect bolt holes.
 - (b) Visually inspect bolts for elongation.

ACCESS/LOCATION

N/A

DETECTABLE CRACK SIZE

Tailcone

INSPECTION PROCEDURE

Visual and Dye Penetrant

REPAIR/MODIFICATION

If cracks or bolt holes are elongated, refer to SE72-3.

COMMENTS

Ref: SE72-3



Vertical Stabilizer Attachment Areas Figure 3-13 (Sheet 1) 1731C1001

TITLE Rudder Stop Clip Inspection

MODEL

EFFECTIVITY

182Q F182Q 18266591 Thru 18267715 F18200026 Thru F18200169

INSPECTION COMPLIANCE

INITIAL 1000 HRS or 3 YRS REPEAT 1000 HRS or 3 YRS Thereafter

PURPOSE

To inspect rudder stop clips for corrosion.

INSPECTION INSTRUCTIONS

Using an inspection mirror, inspect lower surface of rudder stop clip for corrosion.

ACCESS/LOCATION

Elevator torque tube skin slit.

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Replace rudder stop clip.

COMMENTS

Ref: SE80-65

DETECTABLE CRACK SIZE

Aileron Balance Weight Inspection

MODEL

TITLE

EFFECTIVITY

150 Thru 150M	617, 628, 644, 649, 17001 Thru 1507940	5 INSPEC	CTION COMPLIANCE
150K Thru 150M	A1500001 Thru A1500734		
A-150L	A-1501001 Thru A-1501039	NITIAL	3000 HRS
A-A150L	A-A1500001 Thru A-A1500009		
F150F Thru F150M	F150-0001 Thru F15001428	REPEAT	3000 HRS
FA150K Thru FA150M	FA1500001 Thru FA1500336		Thereafter
172 Thru 172Q	610, 612, 615, 622, 625, 630, 638,		
	639, 2800 Thru 17276211		
R172E Thru R172K	R1722000 Thru R1723454		
172RG	691, 172RG0001 Thru 172RG1191		
P172D	P1725720 Thru P17257188		
FP172	FP172-0001 Thru FP172-0003		
F172D Thru F172P	F172-0001 Thru F17202254		
FR172E Thru FR172K	FR17200001 Thru FR17200675		
180 Thru 180K	604, 614, 624, 645, 30000 Thru 1805320	3	
182 Thru 182R	613, 631, 634, 675, 33000 Thru 1826858	6	
R182	R1820001 Thru 18202039		
A182J Thru A182N	A182-0001 Thru A182-0148		
F182P Thru F182Q	F18200001 Thru F18200169		
F182RG	FR18200001 Thru FR18200070		
185 Thru A185F	632, 652, 185-0001 Thru 18504448		

PURPOSE

To inspect aileron balance weight for corrosion and loosen rivets.

INSPECTION INSTRUCTIONS

- 1. Remove aileron in accordance with the applicable Service Manual.
- 2. Visual inspect aileron balance weight for corrosion and loose attach rivet.
- 3. Reinstall aileron in accordance with applicable Service Manual.

ACCESS/LOCATION

DETECTABLE CRACK SIZE

N/A

INSPECTION PROCEDURE

Visual

Aileron

REPAIR/MODIFICATION

Repair corrosion using standard corrosion removal procedures and replace rivets as required. Replace loose rivets with MS204700A5 Rivets.

COMMENTS

Wing Spar Spray Boom Hole Inspection

MODEL

EFFECTIVITY

188B

18801178 Thru 18802057

INSPECTION COMPLIANCE

INITIAL 1000 HRS

REPEAT 1000 HRS Thereafter

PURPOSE

To inspect boom attach holes for elongation.

INSPECTION INSTRUCTIONS

- Remove spray boom and mounting bracket at wing station 103.38 in accordance with Service 1. Manual.
- 2. Using ball gages, inspect the attach holes for elongation.

ACCESS/LOCATION

Lower Wing

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

If holes are elongated greater than 0.210 inch and cone shaped more than half way, modify in accordance with SK188-65.

COMMENTS

Ref: SK188-65

DETECTABLE CRACK SIZE

N/A

TITLE

TITLE Wing Strut and End Fitting

MODEL

EFFECTIVITY

120	8003 Thru 15075	INSPECT	TION COMPLIANCE
140 140A	15200 Thru 15724	INITIAL	12,000 HRS *
1507 1507 A150K Thru A150M A-150L A-A150L F150F Thru F150M FA150K Thru FA150M 152 152A 170 Thru 170B 172 Thru 172Q P172D R172E Thru 172K 172RG FP172 F172D Thru F172P	617, 628, 644, 649, 17001 Thru 15079405 A150001 Thru A1500734 A-1501001 Thru A-1501039 A-A1500001 Thru A-1501039 F150-0001 Thru F15001428 FA1500001 Thru F15001428 FA1500001 Thru F1500336 15279406 Thru 15286033 681, A1500433, A1520735 Thru A1521049 609, 1800 Thru 27169 610, 612, 615, 622, 625, 630, 638, 639, 2800 Thru 17276211 P17257120 Thru P17257188 680, R172-0001 Thru R1723454 172RG0001 Thru F172-0003 F172-0001 Thru FP172-0003 F172-0001 Thru F17202254	REPEAT	2,000 HRS * Thereafter
FR172E Thru FR172K	FR17200001 Thru FR17200675		
113 Thru 1130 180 Thru 180K 182 Thru 182R R182 A182J Thru A182N F182P Thru F1820	604, 614, 624, 645, 30000 Thru 18053203 613, 631, 634, 675, 33000 Thru 18268586 R1820001 Thru 18202039 A182-0001 Thru A182-0148 E18200001 Thru E18200169		
F182RG	FR18200001 Thru FR18200070		
100 THU ATOOF	002,002,100-0001 1110 10004440		

PURPOSE

To check for cracks in wing strut and end fittings.

INSPECTION INSTRUCTIONS

- 1. Visually inspect strut tube for cracks.
- Eddy Current Inspect for cracks radiating from the wing and fuselage attach holes in the wing strut end fittings, using NDT 18 Eddy Current test unit, with 1/2-inch probe. Frequency to be 175. Test Standard shall be Grandia Test Block.

ACCESS/LOCATION

Loosen strut cuff fairing attachments. Support wing and remove strut.

INSPECTION PROCEDURE

Eddy Current and Visual

REPAIR/MODIFICATION

Replace wing strut.

COMMENTS

Report wing strut cracks to Cessna Aircraft Company.

* Should any history of low level overland survey or patrol exist, then the initial inspection compliance is 6000-hours with repeat at 1000-hour intervals thereafter.

DETECTABLE CRACK SIZE



Wing Strut Inspections Figure 3-14 (Sheet 1)

Wing Fuselage Attach Fittings Inspection

EFFECTIVITY

TITLE

120	8003 Thru 15075	INSPECT	ION COMPLIANCE
140	15200 Thru 15724	INITIAL	12 000 HPS
140A	617 629 644 640 17001 Thru 15070405		12,000 003
A150K Thru A150M	A150001 Thru A1500724	DEDEAT	2000 HDC
A 150K THIU A 150W	A 1501001 Thru A 1501020	NEFEAT	Thoroaftor
A-150L	A-A1500001 Thru A-A150000		merealler
E150E Thru E150M	E150-0001 Thru E15001428		
EA150K Thru EA150M	EA1500001 Thru EA1500226		
150 THIU FATSON	15270406 Thru 15286033		
152	621 A1500/22 A1520725 Thru A15210/0		
170 Thru 170B	600 1800 Thru 27160		
172 Thru 1720	610 612 615 622 625 630 638		
172 1110 1720	639 2800 Thru 17276211		
P172D	P17257120 Thru P17257188		
B172E Thru 172K	680 B172-0001 Thru B1723454		
172RG	691 172BG0001 Thru 172BG1191		
FP172	EP172-0001 Thru EP172-0003		
F172D Thru F172P	F172-0001 Thru F17202254		
FR172F Thru FR172K	EB17200001 Thru EB17200675		
175 Thru 175C	619 28700A 55001 Thru 17557119		
180 Thru 180K	604 614 624 645 30000 Thru 18053203		
182 Thru 182B	613, 631, 634, 675, 33000 Thru 18268586		
R182	B1820001 Thru 18202039		2
A182J Thru A182N	A182-0001 Thru A182-0148		
F182P Thru F182Q	F18200001 Thru F18200169		
F182RG	FR18200001 Thru FR18200070		
185 Thru A185F	632, 652, 185-0001 Thru 18504448		

PURPOSE

To eddy current inspect primary load path of forward and aft spar fitting. There is one known history of cracking, except for rear spar web referred to in CAP Item 57-10-04.

INSPECTION INSTRUCTIONS

- 1. Inspect for possible cracking from wing attachment holes outward. (Refer to step 4.)
- 2. Possible galling or deformation of wing attachment holes. (Refer to step 4.)
- 3. Check wing attach bolts for looseness, If blots are loose, remove wing attach bolts and inspect for cracks. (Refer to step 4.)
- 4. Eddy Current Inspect, per instructions in section 4-0, Basic Information, pages x and xi.

ACCESS/LOCATION

Remove wing-to-fuselage fairing strips.

INSPECTION PROCEDURE

Eddy Current

REPAIR/MODIFICATION

Replace cracked parts.

COMMENTS

Contact Cessna Aircraft Company for instructions if oversize holes are found.

DETECTABLE CRACK SIZE







Wing Fuselage Attach Fitting Figure 3-15 (Sheet 1)

Wing Fuselage Attach Fittings Rear Spar Center Lug Inspection

EFFECTIVITY

TITLE

177 Thru 177B 177RG F177RG 190/195/195A/195B 661, 17700001 Thru 17702752 177RG0001 Thru 177RG1366 F177RG0001 Thru F177RG0177 7004 Thru 16183

INSPECTION COMPLIANCE

INITIAL 12,000 HRS

REPEAT 3000 HRS Thereafter

PURPOSE

To eddy current inspect primary load path of forward and aft spar fitting.

INSPECTION INSTRUCTIONS

- 1. Inspect for possible cracking from wing attachment holes outward. (Refer to step 4.)
- 2. Possible galling or deformation of wing attachment holes. (Refer to step 4.)
- 3. Do not remove wing attach bolts unless cracking or looseness is suspected (Refer to step 4.)
- 4. Eddy Current Inspet, per instructions in section 4-0, Basic Information, pages x and xi.

ACCESS/LOCATION

Remove wing-to-fuselage fairing strips.

DETECTABLE CRACK SIZE

N/A

INSPECTION PROCEDURE

Eddy Current

REPAIR/MODIFICATION

Replace cracked parts.

COMMENTS

Contact Cessna Aircraft Company for instructions if oversize holes are found.





DETAIL A

1722C1002 1722X1001

Wing Fuselage Attach Fitting Rear Spar Center Lug Figure 3-16 (Sheet 1)

TITLE Wing Spar Inspection

MODEL

EFFECTIVITY

188 Thru 188A A-A188B 188-0001 Thru 18800832 A-A1880001 Thru A-A18803

INSPECTION COMPLIANCE

INITIAL 1000 HRS

REPEAT 1000 HRS Thereafter

PURPOSE

To inspect for cracks at forward and aft attachment fitting.

INSPECTION INSTRUCTIONS

- 1. Remove wing gap fairing from wing in accordance with Service Manual.
- 2. Using an inspection mirror, inspect angle in the bays inboard and outboard of station 100, giving close attention to surface and aft edge of angle.
- 3. Inspect for hair line cracks around rivets.
- 4. Inspect for cracks in lower flange of spar channel and splice channel.
- 5. Using an inspection mirror, inspect front spar radius area for cracks.
 - a. Give close attention to web radius immediately below and outboard of attachment hole.
 - b. Crack will appear as a hair line extending from web radius toward rivet holes through spar web rib and spar reinforcement immediately outboard of attachment hole.
 - c. Inspect both sides of rear spar for cracks.
- 6. Using an inspection mirror, inspect rear spar radius area for cracks.
 - a. Give close attention to web radius immediately below and outboard of attachment hole.
 - b. Crack will appear as a hair line extending from web radius toward rivet holes through spar web rib and spar reinforcement immediately outboard of attachment hole.
 - c. Inspect both sides of rear spar, If cracks extend beyond rivet, install rear splice in accordance with SK188-30. If not, stop drill crack using 0.098-inch diameter drill.

ACCESS/LOCATION

Wing to Fuselage

INSPECTION PROCEDURE

Visual and Dye Penetrant

REPAIR/MODIFICATION

If cracks extend beyond rivet, install rear splice in accordance with SK188-23A. If not, stop drill crack using 0.098-inch diameter drill.

COMMENTS

Ref: SK188-23A

DETECTABLE CRACK SIZE

N/A

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Front Spar Lower Cap Figure 3-17 (Sheet 1)



Front Spar to Fuselage Attachment Figure 3-17 (Sheet 2)

TITLE Flap Inspection

MODEL SERIES	SERIAL NUMBERS		
140	8001	thru	15075
140A	15200	thru	15724
170	1800	thru	20999
170	25000	thru	27169
170	609		
190/195	7004	thru	7999
190/195	16000	thru	16183

INSPECTION COMPLIANCE

INITIAL 100 hours of operation or 12 months, whichever occurs first.

REPEAT Every 100 hours of operation or 12 months, whichever occurs first.

PURPOSE

To visually inspect the control surface skins for cracks.

INSPECTION INSTRUCTIONS

Visually inspect each control surface skin for cracks. Carefully inspect the bend radius areas of corrugated skins for cracks.

ACCESS/LOCATION

Flap

INSPECTION PROCEDURE

Visual

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REPAIR/MODIFICATION

Any smooth dents in the skin that are free from cracks, abrasions, and sharp corners, which are not stress wrinkles, and do not interfere with any internal structure or mechanism, may be considered as negligible damage. In areas of low stress intensity, deep scratches, or deep sharp dents, which, after trimming or stop drilling, can be enclosed by a two inch circle, can be considered negligible, if the damaged area is at least one diameter of the enclosed circle away from all exterior rivet lines and material edges. Damage that is not negligible, shall be repaired prior to returning airplane to service.

Cracks in corrugated skins;

- 1. It is permissible to stop drill a crack that originates at the trailing edge of the control surface or from a trailing edge rivet and is not more than 2 inches in length.
- 2. Stop drill crack using a drill #30 (.128 inch) drill.
- 3. A crack may only be stop drilled once.

NOTE: A crack that passes through a flap trailing edge rivet and does not extend to the trailing edge of the skin may be stop drilled at both ends of the crack.

- 4. Any control surface that has a crack that progresses past a stop drilled hole shall have an approved repair or skin replacement made as soon as practicable.
- 5. A control surface that has any of the following conditions shall have an approved repair or skin replacement made as soon as practicable:

A crack that is longer than 2 inches.

- B. A crack that does not originate from the trailing edge or a trailing edge rivet.
- C. Cracks in more than six trailing edge rivet locations per skin.
- 6. Affected control surfaces with corrugated skins and having a stop drilled crack that does not extend past the stop drill hole, may remain in service without additional repair.

COMMENTS

Temporary Revision Number 2 7 January 2000



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Flap Inspection Figure 3-18 (Sheet 1)

TITLE

Flap Support Bracket, Flap Track, and Roller Inspection

MODEL

EFFECTIVITY

8003 Thru 15075	INSPEC	CTION CO	MPLIANCE
8001 Thru 15075			
15200 Thru 15724		INITIAL	1000 HRS
617, 628, 644, 649, 17001 Thru 1503	79405		
A150001 Thru A1500734		REPEAT	500 HRS
A-1501001 Thru A-1501039			Thereafter
A-A1500001 Thru A-A1500009			
F150-0001 Thru F15001428			
FA1500001 Thru FA1500336			
15279406 Thru 15286033			
681, A1500433, A1520735 Thru A15	21049		
609, 1800 Thru 27169			
610, 612, 615, 622, 625, 630, 638,			
639, 2800 Thru 17276211			
P17257120 Thru P17257188			
FP172-0001 Thru FP172-0003			
F172-0001 Thru F17202254			
FR17200001 Thru FR17200675			
619, 28700A, 55001 Thru 17557119			
604, 614, 624, 645, 30000 Thru 1805	53203		
613, 631, 634, 675, 33000 Thru 1820	58586		
R1820001 Thru 18202039			
A182-0001 Thru A182-0148			
F18200001 Thru F18200169			
FR18200001 Thru FR18200070			
632, 652, 185-0001 Thru 18504448			
	8003 Thru 15075 8001 Thru 15075 15200 Thru 15724 617, 628, 644, 649, 17001 Thru 150 A150001 Thru A1500734 A-1501001 Thru A-1501039 A-A1500001 Thru A-1500009 F150-0001 Thru F15001428 FA1500001 Thru F15001428 FA1500001 Thru F1500336 15279406 Thru 15286033 681, A1500433, A1520735 Thru A15 609, 1800 Thru 27169 610, 612, 615, 622, 625, 630, 638, 639, 2800 Thru 17276211 P17257120 Thru P17257188 FP172-0001 Thru FP172-0003 F172-0001 Thru FP172-0003 F172-0001 Thru FR17200675 619, 28700A, 55001 Thru 17557119 604, 614, 624, 645, 30000 Thru 1808 613, 631, 634, 675, 33000 Thru 1808 613, 631, 634, 675, 33000 Thru 1826 R1820001 Thru A182-0148 F18200001 Thru F18200169 FR18200001 Thru FR18200070 632, 652, 185-0001 Thru 18504448	8003 Thru 15075 INSPEC 8001 Thru 15075 15200 Thru 15724 617, 628, 644, 649, 17001 Thru 15079405 A150001 Thru A1500734 A-1501001 Thru A-1501039 A-A1500001 Thru A-A1500009 F150-0001 Thru F15001428 FA1500001 Thru F15001428 F09, 1800 Thru 15286033 681, A1500433, A1520735 Thru A1521049 609, 1800 Thru 27169 610, 612, 615, 622, 625, 630, 638, 639, 2800 Thru 17276211 P17257120 Thru P17257188 FP172-0001 Thru F17202254 FR1720001 Thru F17200675 619, 28700A, 55001 Thru 17557119 604, 614, 624, 645, 30000 Thru 18053203 613, 631, 634, 675, 33000 Thru 18268586 R1820001 Thru A182-0148 F18200001 Thru F18200169 FR18200001 Thru FR18200070 632, 652, 185-0001 Thru 18504448	8003 Thru 15075 INSPECTION CC 8001 Thru 15075 INITIAL 617, 628, 644, 649, 17001 Thru 15079405 A150001 Thru A1500734 REPEAT A-1501001 Thru A-1501039 A-A1500001 Thru A-A1500009 F150-0001 Thru F15001428 FA1500001 Thru F15001428 FA1500001 Thru F15001428 FA1500001 Thru F15001428 FA1500001 Thru F15001428 FA1500001 Thru F286033 681, A1500433, A1520735 Thru A1521049 609, 1800 Thru 27169 610, 612, 615, 622, 625, 630, 638, 639, 2800 Thru 17276211 P17257120 Thru P17257188 FP172-0001 Thru F17200675 619, 28700A, 55001 Thru 17557119 604, 614, 624, 645, 30000 Thru 18053203 613, 631, 634, 675, 33000 Thru 18268586 R1820001 Thru A182-0148 F18200001 Thru F18200169 FR18200001 Thru FR18200070 632, 652, 185-0001 Thru 18504448 S0001 Thru 18504448

PURPOSE

To ensure flap support bracket flap track and roller integrity.

INSPECTION INSTRUCTIONS

- 1. Visually inspect flap support bracket for cracks.
- 2. Visually inspect flap track, rollers and bushings for damage and excessive wear

ACCESS/LOCATION

Lower wing flap area.

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Replace flap support bracket, flap track, rollers and bushings as required.

COMMENTS

DETECTABLE CRACK SIZE

TITLE Aileron Inspection

SERIAL NUMBERS			IODEL SERIES		
15075	thru	8003		120	12
15075	thru	8001		140	14
15724	thru	15200		140A	14
20999	thru	1800		170	17
27169	thru	25000		170	17
		609		170	17
7999	thru	7004		190/195	19
16183	thru	16000		190/195	19
7999 16183	thru thru	7004 16000		190/195 190/195	19 19

INSPECTION COMPLIANCE

INITIAL 100 hours of operation or 12 months, whichever occurs first.

REPEAT Every 100 hours of operation or 12 months, whichever occurs first.

PURPOSE

N

To visually inspect the control surface skins for cracks.

INSPECTION INSTRUCTIONS

Visually inspect each control surface skin for cracks. Carefully inspect the bend radius areas of corrugated skins for cracks.

ACCESS/LOCATION

Aileron

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Any smooth dents in the skin that are free from cracks, abrasions, and sharp corners, which are not stress wrinkles, and do not interfere with any internal structure or mechanism, may be considered as negligible damage. In areas of low stress intensity, deep scratches, or deep sharp dents, which, after trimming or stop drilling, can be enclosed by a two inch circle, can be considered negligible, if the damaged area is at least one diameter of the enclosed circle away from all exterior rivet lines and material edges. Damage that is not negligible, shall be repaired prior to returning airplane to service.

Cracks in corrugated skins;

- 1. It is permissible to stop drill a crack that originates at the trailing edge of the control surface or from a trailing edge rivet and is not more than 2 inches in length.
- 2. Stop drill crack using a #30 (.128 inch) drill.
- 3. A crack may only be stop drilled once.

NOTE: A crack that passes through a flap trailing edge rivet and does not extend to the trailing edge of the skin may be stop drilled at both ends of the crack.

- 4. Any control surface that has a crack that progresses past a stop drilled hole shall have an approved repair or skin replacement made as soon as practicable.
- 5. A control surface that has any of the following conditions shall have an approved repair or skin replacement made as soon as practicable:
 - A. A crack that is longer than 2 inches.
 - B. A crack that does not originate from the trailing edge or a trailing edge rivet.
 - C. Cracks in more than six trailing edge rivet locations per skin.
- 6. Affected control surfaces with corrugated skins and having a stop drilled crack that does not extend past the stop drill hole, may remain in service without additional repair.

COMMENTS

Temporary Revision Number 2 7 January 2000



Ε

Aileron Inspection Figure 3-19 (Sheet 1)

TITLE

Engine Mount Welded Joint Inspection

MODEL

EFFECTIVITY

120 140 140A 150 Thru 150M A150K Thru A150M A-150L A-A150L F150F Thru F150M FA150K Thru FA150M 152 A152 F152 FA152 170 Thru 170B 172 Thru 172Q P172D R172E Thru 172K 172RG FP172 F172D Thru F172P FR172E Thru FR172K 177 Thru 177B 177RG F177RG 175 Thru 175C 180 Thru 180K 182 Thru 182R R182 A182J Thru A182N F182P Thru F182Q F182RG 188 Thru T188C A-A188B 190/195/195A/195B

8003 Thru 15075 8001 Thru 15075 15200 Thru 15724 617, 628, 644, 649, 17001 Thru 15079405 A150001 Thru A1500734 A-1501001 Thru A-1501039 A-A1500001 Thru A-A1500009 F150-0001 Thru F15001428 FA1500001 Thru FA1500336 15279406 Thru 15286033 681, A1500433, A1520735 Thru A1521049 F15201429 Thru F15201980 FA1520337 Thru FA1520425 609, 1800 Thru 27169 610, 612, 615, 622, 625, 630, 638, 639, 2800 Thru 17276211 P17257120 Thru P17257188 680, R172-0001 Thru R1723454 691, 172RG0001 Thru 172RG1191 FP172-0001 Thru FP172-0003 F172-0001 Thru F17202254 FR17200001 Thru FR17200675 661, 17700001 Thru 17702752 177RG0001 Thru 177RG1366 F177RG0001 Thru F177RG0177 619, 28700A, 55001 Thru 17557119 604, 614, 624, 645, 30000 Thru 18053203 613, 631, 634, 675, 33000 Thru 18268586 R1820001 Thru 18202039 A182-0001 Thru A182-0148 F18200001 Thru F18200169 FR18200001 Thru FR18200070 653, 678T, 188-0001 Thru T18803974T A-A1880001 Thru A-A188034 7004 Thru 16183

PURPOSE

To inspect welded joints on the engine mounts for cracks.

INSPECTION INSTRUCTIONS

Dye Penetrant inspect all welded joints on the engine mounts.

ACCESS/LOCATION

Engine Nacelle

INSPECTION PROCEDURE

Visual and Dye Penetrant

INSPECTION COMPLIANCE

INITIAL 10,000 HRS

REPEAT 5000 HRS Thereafter

DETECTABLE CRACK SIZE

REPAIR/MODIFICATION

If cracks are found, contact Cessna Aircraft Company and describe condition.

COMMENTS



Engine Mount Welded Joint Inspection Figure 3-20 (Sheet 1)

TITLE Inspect Engine Mount For Corrosion

MODEL

EFFECTIVITY

604, 614, 624, 645,

613, 631, 634, 675,

30000 Thru 18053203

33000 Thru 18268586

R1820001 Thru 18202039

A182-0001 Thru A182-0148

F18200001 Thru F18200169

FR18200001 Thru FR18200070

180 Thru 180K

182 Thru 182R

R182 A182J Thru A182N F182P Thru F182Q F182RG

PURPOSE

To inspect engine mounts for Corrosion.

INSPECTION INSTRUCTIONS

Visually inspect engine mount for corrosion.

ACCESS/LOCATION

Engine Nacelle

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Remove corrosion using standard corrosion removal procedures, if corrosion is greater than 10% of the wall thickness replace engine mount.

COMMENTS

INSPECTION COMPLIANCE

INITIAL 10,000 HRS or 5 YRS REPEAT 5000 HRS or 3 YRS Thereafter

DETECTABLE CRACK SIZE

Cowl Flap Hinge Pin Inspection

MODEL

TITLE

EFFECTIVITY

180 Thru 180K 182 Thru 182R R182 A182J Thru A182N F182P Thru F182Q F182RG 185 Thru A185F 604, 614, 624, 645, 30000 Thru 18053203 613, 631, 634, 675, 33000 Thru 18268586 R1820001 Thru 18202039 A182-0001 Thru A182-0148 F18200001 Thru F1820069 FR18200001 Thru FR18200070 632, 652, 185-0001 Thru 18504448

INSPECTION COMPLIANCE

INITIAL 1000 HRS or 2 YRS REPEAT 1000 HRS or 2 YRS

PURPOSE

To inspect cowl flap for security and wear.

INSPECTION INSTRUCTIONS

- 1. Inspect cowl flap hinges for security and wear.
- 2. Check hinge pin for wear and fit in hinge.
- 3. Ensure hinge pin is safety wired in position.

ACCESS/LOCATION

Bottom of Nacelle

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Repair and replace hinge and hinge pin.

COMMENTS

Ref: SE71-27

DETECTABLE CRACK SIZE

TITLE Oil Cooler Inspection

MODEL

EFFECTIVITY

152	15279406 Thru 15286033	INSPECT	FION COMPLIANCE
A152	A1520735 Thru A1521049		•
F152	F15201429 Thru F15201980	INITIAL	1000 HRS
FA152	FA1520337 Thru FA1520425		or 3 YRS
172 Thru 172Q	610, 612, 615, 622, 625, 630, 638,	REPEAT	1000 HRS
	639, 2800 Thru 17276211		or 3 YRS
P172D	P17257120 Thru P17257188		Thereafter
172RG	691, 172RG0001 Thru 172RG1191		
FP172	FP172-0001 Thru FP172-0003		
F172D Thru F172P	F172-0001 Thru F17202254		
FR172E Thru FR172K	FR17200001 Thru FR17200675		
177 Thru 177B	661, 17700001 Thru 17702752		
177RG	177RG0001 Thru 177RG1366		
F177RG	F177RG0001 Thru F177RG0177		
182 Thru 182R	613, 631, 634, 675, 33000 Thru 18268586		
R182	R1820001 Thru 18202039		
F182P Thru F182Q	F18200001 Thru F18200169		
F182RG	FR18200001 Thru FR18200070		
A182J Thru A182N	A182-0001 Thru A182-0148		

PURPOSE

To inspect oil cooler for internal corrosion.

INSPECTION INSTRUCTIONS

Inspect oil cooler for corrosion. (Refer to SE80-96.)

ACCESS/LOCATION

Engine Nacelle

INSPECTION PROCEDURE

Visual

REPAIR/MODIFICATION

Repair in accordance with SE80-96.

COMMENTS

Ref: SE80-96

DETECTABLE CRACK SIZE